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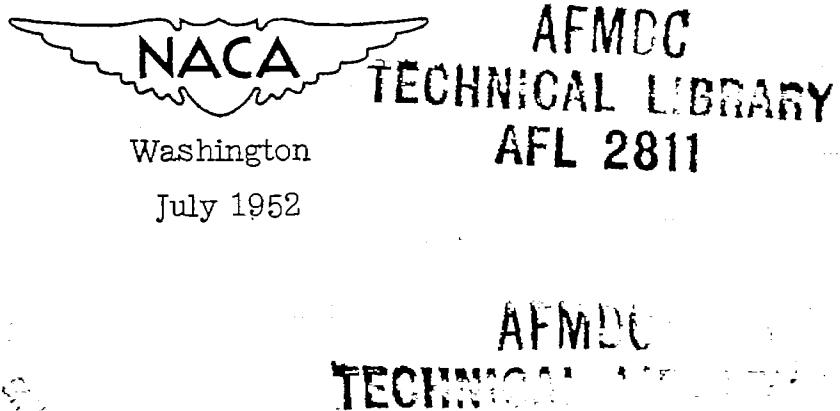
# NATIONAL ADVISORY COMMITTEE FOR AERONAUTICS

TECHNICAL NOTE 2744

## PRACTICAL CALCULATION OF SECOND-ORDER SUPERSONIC FLOW PAST NONLIFTING BODIES OF REVOLUTION

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## SUMMARY

Calculation of second-order supersonic flow past bodies of revolution at zero angle of attack is described in detail, and reduced to routine computation. Use of an approximate tangency condition is shown to increase the accuracy for bodies with corners. Tables of basic functions and standard computing forms are presented. The procedure is summarized so that one can apply it without necessarily understanding the details of the theory. A sample calculation is given, and several examples are compared with solutions calculated by the method of characteristics.

## INTRODUCTION

For predicting the pressure distribution over a nonlifting body of revolution in supersonic flow, linearized theory is often found to be inadequate. In the past, greater accuracy could be achieved only by resorting to the laborious method of characteristics. Recently, however, a second-order solution has been found which within its range of applicability yields greater accuracy than linearized theory, while requiring considerably less labor than the method of characteristics.

The present paper aims to give a complete description of the second-order method, and to reduce it to routine computation. Previously published descriptions of the procedure, which are inadequate in some respects, are revised. Shortcuts in the computing scheme are pointed out. Extensive tables of the required basic solutions are presented, to be used in conjunction with standard computing forms. Several examples illustrate the procedure.

The reader interested only in calculating the second-order solution for a definite body, without necessarily understanding the details of

the theory, can turn directly to the final section Practical Use of Method on page 26.

### NOTATION

$a, b, c \}$	functions of $t$ associated with linear and quadratic
$d, e, f \}$	source solutions
$g, h, i, j \}$	functions of $t$ associated with step, corner, and curva-
$k, l, m \}$	ture solutions
$c_p$	pressure coefficient
$E$	complete elliptic integral of second kind with modulus $k = \sqrt{(1-t)/(1+t)}$
$G_0$	function associated with determination of first interval
$G_1$	function associated with determination of subsequent intervals
$K$	complete elliptic integral of first kind with modulus $k = \sqrt{(1-t)/(1+t)}$
$M$	free-stream Mach number
$N$	$\frac{\gamma+1}{2} \frac{M^2}{\beta^2}$
$P_n$	nth point on surface of body
$q$	resultant velocity
$r$	radial coordinate
$R$	local radius of body
$S(x)$	source strength distribution function
$t$	conical variable $\left( \frac{\beta r}{x} \right)$

u	axial velocity component
v	radial velocity component
x	axial coordinate
$\beta$	$\sqrt{M^2 - 1}$
$\gamma$	adiabatic exponent of gas
$\delta_n$	length of interval between points $P_n$ and $P_{n+1}$
$\varphi$	first-order (linearized) perturbation potential
$\varphi^{(m)}$	basic first-order solution homogeneous of order $m$
$\psi$	second-order perturbation potential
$\Phi$	exact perturbation potential
X	complementary function for second-order solution
$\Psi$	particular integral for second-order solution

## Superscripts

(1)	first-order value
(2)	second-order value
	differentiation with respect to $x$

## Subscripts

o	value at tip of pointed body
n	value at nth point on body, $P_n$
c	value at corner

## DETAILS OF SECOND-ORDER SOLUTION

The natural way of attempting to improve a first-order (linearized) solution is by iteration. For nonlifting bodies of revolution, the second-order iteration equation was solved in principle in 1949 by the discovery of a particular integral expressed in terms of the first-order solution (reference 1). This reduces the second-order problem to the form of the first-order problem. For supersonic speeds, both problems can then be solved by suitable modification of the method of Kármán and Moore (reference 2). The result is the axially symmetric counterpart of Busemann's second-order solution for plane supersonic flow (reference 3), to which it reduces locally at a corner.

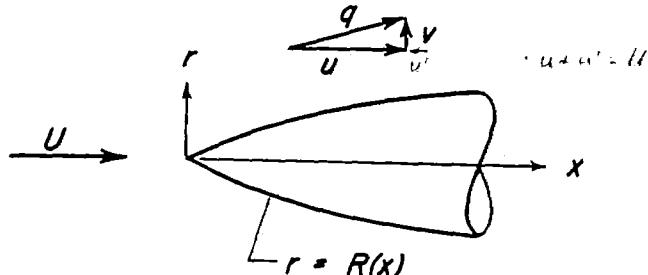
As a preliminary to describing this procedure in detail, the reduction of the second-order problem will be summarized. Further details will be found in references 1 and 4.

## Reduction of Second-Order Problem to Two First-Order Problems

At moderate supersonic speeds, the flow past a reasonably slender body of revolution is nearly isentropic and therefore nearly irrotational. To this approximation, there exists a perturbation potential  $\Phi$  whose derivatives give the velocity perturbations (referred to the velocity  $U$  of the free stream), so that

$$\left. \begin{aligned} \frac{u}{U} &= \frac{U + u'}{U} \\ \frac{v}{U} &= \Phi_r \end{aligned} \right\} \quad (1)$$

Here subscripts indicate differentiation, and the notation is explained by sketch (a). The equations of motion for a polytropic gas combine into the single equation



Sketch (a)

in cylinder coordinates:

$$\Phi_{rr} + \frac{\Phi_r}{r} - \beta^2 \Phi_{xx} = M^2 \left[ \begin{array}{l} 2(N-1)\beta^2 \Phi_x \Phi_{xx} + 2\Phi_r \Phi_{xr} + \\ \Phi_r^2 \Phi_{rr} + \text{other cubic terms} \end{array} \right] \quad \left. \begin{array}{l} \\ \\ \end{array} \right\} (2)$$

where

$$\beta^2 = M^2 - 1$$

$$N = \frac{\gamma+1}{2} \frac{M^2}{\beta^2}$$

Here all linear terms have been grouped on the left and quadratic and cubic terms on the right. The only cubic term which gives a second-order contribution is the one involving  $\Phi_r^2 \Phi_{rr}$ .

This equation must be solved subject to the boundary conditions that all disturbances vanish ahead of the body, and that the flow is tangent to the surface of the body.

Iteration procedure. - The equation of motion (2) cannot be solved directly because it is nonlinear. Therefore a method of successive approximations is adopted - the so-called Prandtl-Busemann iteration procedure.

In the first approximation, the nonlinear right-hand side of equation (2) is neglected altogether. Hence the first-order perturbation potential  $\phi$  satisfies the familiar wave equation of linearized supersonic theory:

$$\Phi_{rr} + \frac{\Phi_r}{r} - \beta^2 \Phi_{xx} = 0 \quad (3)$$

In the second approximation, the right-hand side of equation (2) is no longer entirely neglected but is evaluated approximately in terms of the previously determined first-order solution. Hence the second-order perturbation potential  $\phi$  satisfies the nonhomogeneous wave equation

$$\phi_{rr} + \frac{\phi_r}{r} - \beta^2 \phi_{xx} = M^2 [2(N-1)\beta^2 \Phi_x \Phi_{xx} + 2\Phi_r \Phi_{xr} + \Phi_r^2 \Phi_{rr}] \quad (4)$$

Here  $\phi$  will be taken to be the complete second-order perturbation potential, rather than a correction to the first-order solution.

This procedure could be continued to third and higher approximations, subject to the limitation that at some stage the effects of

entropy variations, which were ignored in assuming potential flow, would exceed the remainder in the iteration procedure. For slender bodies at moderate Mach numbers, Lighthill has shown (reference 5) that this limit is reached only in the sixth approximation. For practical purposes, however, only the first two steps appear to be useful.

Particular integral.- Solution of the second-order problem is greatly simplified by the discovery that a particular integral  $\psi$  of the iteration equation (4) is given in terms of the first-order solution by

$$\psi = M^2 \left[ \varphi_x(\varphi + Nr\varphi_r) - \frac{1}{4} r\varphi_r^3 \right] \quad (5a)$$

so that

$$\left. \begin{aligned} \psi_x &= M^2 \left[ \varphi_{xx}(\varphi + Nr\varphi_r) + \varphi_x(\varphi_x + Nr\varphi_{xr}) - \frac{3}{4} r\varphi_{xr}\varphi_r^2 \right] \\ \psi_r &= M^2 \left\{ \varphi_{xr}(\varphi + Nr\varphi_r) + \varphi_x \left[ (N+1)\varphi_r + Nr\varphi_{rr} \right] - \frac{1}{4} \varphi_r^2(\varphi_r + 3r\varphi_{rr}) \right\} \end{aligned} \right\} \quad (5b)$$

This reduces the second-order problem to the form of the first-order problem, because the nonhomogeneous iteration equation (4) is reduced to the homogeneous equation (3) of first-order theory. The complete second-order potential consists of the particular integral plus a complementary function  $X$  which is required to re-establish the boundary conditions:

$$\phi = \psi + X \quad (6)$$

and  $X$  is a solution of the first-order equation (3). Thus the remaining problem for  $X$  differs from that for the first-order potential  $\varphi$  only in that the tangency condition is more complicated. Methods for solving first-order problems are well established, so that in principle the second-order problem is solved. In practice, however, various details require careful consideration, to which the subsequent discussion is devoted.

#### Tangency Condition

Because approximations were made in the equation of motion, one would anticipate that a corresponding approximation is permissible in the condition of tangent flow at the body. Such an approximation can be made, and it can be shown that the mathematical order of the error is not thereby increased. This suggests that it is immaterial whether or not the approximation is adopted. However, numerical examples show that the

approximation has in some cases a large effect upon the solution, so that the choice of tangency condition must be carefully considered.<sup>1</sup>

Exact and approximate tangency conditions.- If the body is defined by  $r = R(x)$ , the exact tangency condition for the original problem of equation (2) is

$$\frac{d}{dx} \left( \frac{v}{u} \right) \approx \frac{\phi_r}{1+\phi_x}, \quad \Phi_r = R'(1+\phi_x) \quad \text{at } r = R(x) \quad (7)$$

where the prime indicates differentiation with respect to  $x$ . The corresponding exact tangency conditions for the first- and second-order problems of equations (3) and (4) are

$$\Phi_r = R'(1+\phi_x) \quad \text{at } r = R(x) \quad (8)$$

and

$$\phi_r = R'(1+\phi_x) \quad \text{at } r = R(x) \quad (9)$$

*exact*

(9)

Now in equation (8) it is consistent with the approximations of the first-order theory to neglect the small quantity  $\phi_x$  in comparison with unity. Thus the approximate first-order tangency condition becomes

$$\Phi_r = R' \quad \text{at } r = R(x) \quad (10)$$

Similarly, in equation (9) the term  $\phi_x$  can be replaced by its first-order counterpart. Thus the approximate second-order tangency condition becomes

$$\phi_r = R'(1+\phi_x) \quad \text{at } r = R(x) \quad (11a)$$

or, separating the second-order term into particular integral and complementary function according to equation (6) and collecting known quantities on the right-hand side,

$$\chi_r = R'(1+\phi_x) - \psi_r \quad \text{at } r = R(x) \quad (11b)$$

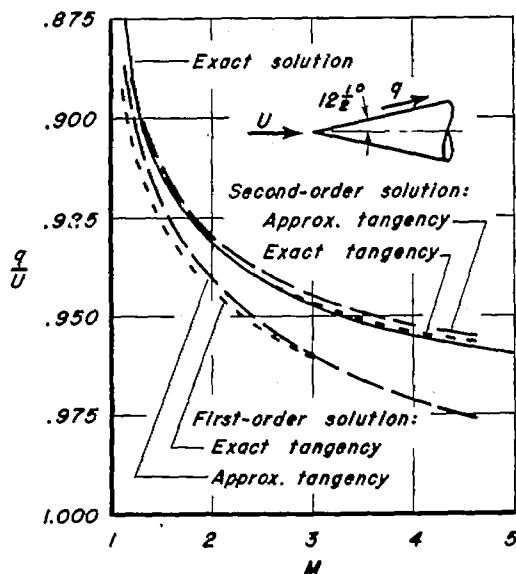
X

Smooth bodies.- For bodies without corners, the choice of tangency condition has no consistent effect upon the error in surface velocity. Greater accuracy in the second-order solution results from using the exact tangency condition in some cases, but the approximate condition

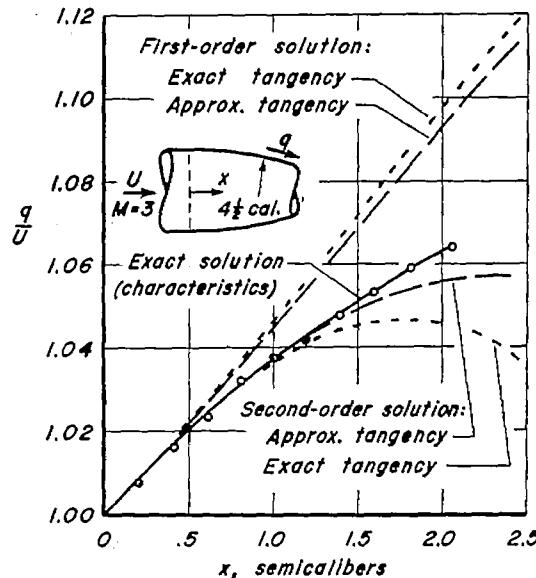
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<sup>1</sup>The magnitude of this effect was brought to the author's attention by John Huth and E. P. Williams of the Rand Corporation.

in others.<sup>2</sup> For example, the exact condition leads to greater accuracy for cones, as shown in sketch (b). This superiority, of course, arises at the tip of any pointed body and persists for some distance downstream. On the other hand, the approximate tangency condition leads to greater accuracy for the boattail following a long cylinder shown in sketch (c), for which the exact solution has been determined by the



Sketch (b)



Sketch (c)

method of characteristics. Thus the conclusion, based upon estimates of the order of error, that neither tangency condition is consistently more accurate, is confirmed empirically for smooth bodies.

Bodies with corners.— In plane flow, the approximate tangency condition invariably leads to more accurate first- and second-order velocities than the exact condition. The superiority of the approximate tangency condition is most pronounced for expansions, and becomes greater as the Mach number falls toward unity.

At a corner on a body of revolution the flow is locally two-dimensional. Therefore the approximate tangency condition is, at least locally, consistently superior to the exact condition for both the

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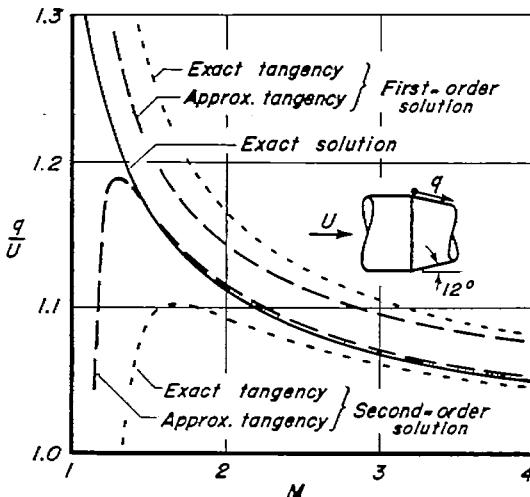
<sup>2</sup>In the first-order solution, however, the approximate tangency condition seems invariably to yield greater accuracy.

first- and second-order solutions. This is shown in sketch (d) for the velocity just behind the corner of a conical boattail which follows a very long circular cylinder. (The exact solution is, of course, given by a plane Prandtl-Meyer expansion.) At moderate Mach numbers, the superiority of the approximate tangency condition is of considerable practical importance in the second-order solution. The superiority is not confined to the immediate vicinity of the corner, but persists far downstream. This is illustrated in sketch (e) by comparison with the solution for a conical boattail calculated by the method of characteristics. (For clarity, the first-order solutions are only partially shown.)

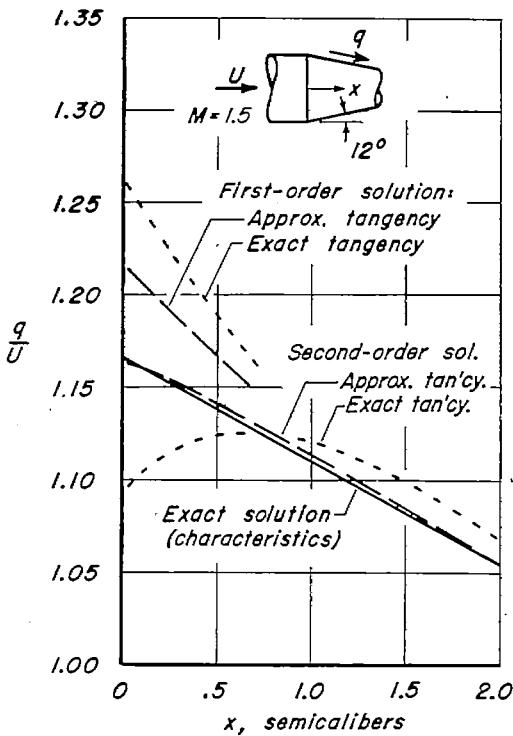
Sketch (d) suggests that the large discrepancy associated with the choice of tangency condition is in some sense a transonic phenomenon. This is confirmed by examination of the expressions for the streamwise velocity just behind the corner. For expansion through an angle whose tangent is  $\epsilon$ , the second-order solution using the exact tangency condition is

$$\frac{u}{U} = 1 + \frac{\epsilon}{\beta - \epsilon} - \frac{\gamma+1}{4} \frac{M^4}{\beta} \frac{\epsilon^2}{(\beta - \epsilon)^3}$$

(12a)



Sketch (d)



Sketch (e)

whereas the second-order solution using the approximate tangency condition is

$$\frac{u}{U} = 1 + \frac{\epsilon}{\beta} + \frac{\epsilon^2}{\beta^2} - \frac{\gamma+1}{4} \frac{M^4}{\beta^4} \epsilon^2 \quad (12b)$$

The difference between these two results is clearly of order  $\epsilon^3$  and hence of third order in the usual sense, according to which linearized theory gives the first approximation. However, in the transonic range (where  $\beta$  is of order  $\epsilon^{1/3}$  for small disturbances) the main term in the difference is

$$\frac{\Delta u}{U} \sim \frac{3(\gamma+1)}{4} \frac{M^4}{\beta^5} \epsilon^3 \quad (12c)$$

which is small only of order  $\epsilon^{4/3}$ . Since  $u/U$  itself is of order  $\epsilon^{2/3}$  in the transonic range, it is seen that the discrepancy has grown to be of second order in the sense of transonic small-disturbance theory. This is simply another example of the fact, which plagues all users of transonic small-disturbance theory, that higher-order effects are greater in the transonic range than at other speeds.

Choice of tangency condition. - It has been seen that although for smooth bodies neither tangency condition can be preferred, for bodies with corners the approximate condition is consistently superior to the exact condition in both first and second order. Consequently, the approximate tangency condition (equations (10) and (11)) is adopted for use henceforth.<sup>3</sup>

The approximate tangency condition has several minor additional advantages. As might be expected, the computing procedure is simplified. For example, the second-order velocities on the surface of a cone, which could not conveniently be written in explicit form in reference 1 (where the exact tangency condition was used) are not unduly complicated if the approximate condition is used. The result is that

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<sup>3</sup>All numerical examples given in references 1 and 4 were calculated using the exact tangency condition, and will therefore not agree precisely with results from the present computing scheme. It should also be noted that the solution presented in references 1 and 4 for the 3-1/2-caliber-long ogive at  $M = 3.24$  is inaccurate near the nose because linear rather than quadratic source solutions were used for calculating the complementary function  $X$ , which results in appreciable error where the body slope is nearly that of the Mach cone.

at the surface of a cone of semivertex angle  $\tan^{-1} \epsilon$

$$\frac{u}{U} = 1 - \epsilon^2 \frac{\operatorname{sech}^{-1} T}{\sqrt{1-T^2}} + \epsilon^4 \left( \frac{\operatorname{sech}^{-1} T}{\sqrt{1-T^2}} \right)^2 + \frac{M^2 \epsilon^4}{1-T^2} \left[ -(\operatorname{sech}^{-1} T)^2 + \frac{10+T^2}{4} \frac{\operatorname{sech}^{-1} T}{\sqrt{1-T^2}} - \left( N + \frac{7}{4} \right) + (N-1)T^2 \left( \frac{\operatorname{sech}^{-1} T}{\sqrt{1-T^2}} \right)^2 \right] \quad (13a)$$

$$\frac{v}{U} = \epsilon \left( 1 - \epsilon^2 \frac{\operatorname{sech}^{-1} T}{\sqrt{1-T^2}} \right) \quad (13b)$$

where  $T = \beta \epsilon$ .

Another advantage is that with the approximate tangency condition the first-order solution exactly satisfies the supersonic similarity rule (the supersonic counterpart of the Göthert rule, reference 6).

#### Pressure Relation

After the velocity components are determined, the pressure coefficient is given by

$$C_p = \frac{2}{\gamma M^2} \left[ \left\{ 1 + \frac{\gamma-1}{2} M^2 \left[ 1 - (1+\Phi_x)^2 - \Phi_r^2 \right] \right\}^{\frac{\gamma}{\gamma-1}} - 1 \right] \quad (14)$$

It was shown in reference 4 that approximating this expression by the leading terms of its series expansion cannot generally be justified, and numerical examples show that such expansion leads to unnecessary loss of accuracy, particularly in the second-order solution (references 1 and 4). Therefore the complete pressure relation of equation (14) is used in the present computing scheme.

#### Basic Solutions of First-Order Equation

It has been seen that discovery of a particular integral reduces the second-order problem to a sequence of two first-order problems. These are best solved by repeated superposition of five basic solutions, which are derived and tabulated below.

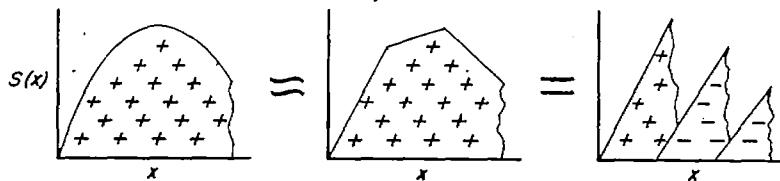
Any first-order solution may be regarded as resulting from a continuous distribution of supersonic sources along the axis of the body.

(See, for example, reference 2 or 7.) A source distribution of local strength  $S(x)$  per unit length yields a first-order perturbation potential given by  $\phi(x, r)$

$$\phi(x, r) = - \int_{-\infty}^{x - \beta r} \frac{S(\xi) d\xi}{\sqrt{(x - \xi)^2 - \beta^2 r^2}} \quad (15)$$

Therefore the first-order problem consists simply in determining the source-distribution function  $S(x)$  which produces the desired shape. However, substituting this expression into the tangency condition yields an integral equation which cannot be solved exactly.

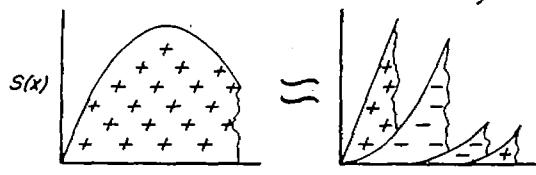
The Kármán-Moore procedure for obtaining an approximate numerical solution involves the assumption that the unknown source function  $S(x)$  can be replaced by a broken line, as indicated in sketch (f). Another



Sketch (f)

(quite equivalent) viewpoint is that the function is approximated by the sum of a number of linear source distributions having various starting points, as shown. The slope of each of these linear elements is determined in succession by imposing the tangency condition at corresponding points along the body. (The details of this procedure are clearly described in Sauer's book, reference 7.)

For calculating a first-order solution which forms the first step of a second-order solution, this broken-line approximation to the source strength is too crude. Although the final second-order velocities are given by first derivatives of  $\phi$ , they involve second derivatives of the first-order solution  $\phi$ , which enter through the particular integral. (See equations (5a) and (5b).) Since differentiation is a roughening process, this means that the first-order potential must be one degree smoother when used as the basis for a second-order solution. This is achieved by approximating the unknown source strength by quadratic rather than linear elements, as shown in sketch (g). However, as

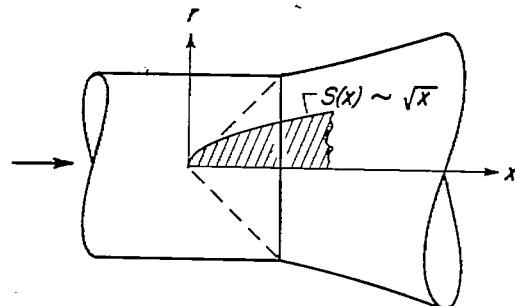


Sketch (g)

indicated in the sketch, the linear element is also required for use at the tip of a pointed body, where the source strength actually rises linearly.

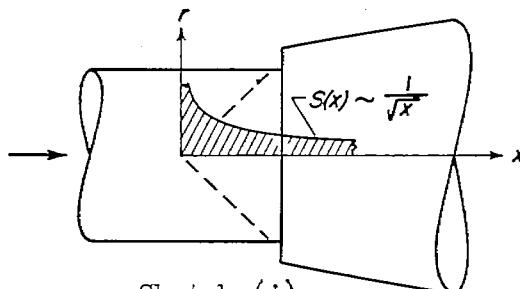
For a smooth body with continuous curvature these two basic solutions are sufficient. Others are required, however, if the body has corners or discontinuities in curvature, which require special treatment. A corner

is accounted for in the first-order solution by adding a source distribution of square-root strength, which produces a discontinuity in streamline slope along its foremost Mach cone. As indicated in sketch (h), this corner solution must be shifted upstream so that its effect first reaches the surface just at the corner. In the same way, a curvature discontinuity is accounted for in the first-order solution by adding a source distribution of  $3/2$ -power strength, which produces a discontinuity in streamline curvature along its foremost Mach cone. This curvature solution is required also at a corner, because an apparent curvature discontinuity remains after the corner solution is added.



Sketch (h)

Because of the roughening due to differentiation, the particular integral has stronger discontinuities than the first-order solution. Thus in the case of a discontinuity in body curvature the particular integral behaves like a corner solution, while in the case of an actual corner it behaves like the solution at a step in the streamlines (sketch (i)). These spurious discontinuities must be canceled in the complementary function. For this purpose the corner solution is used again in the first case. In the second case, another basic solution is required which produces an actual step in the streamlines. As indicated in sketch (i), this step solution results from an inverse square-root source distribution.



Sketch (i)

To summarize, the first-order solution and complementary function are calculated by superposing the following five basic solutions:

1. Linear source solution - used at tip of pointed body
2. Quadratic source solution - used thereafter for body having continuous curvature
3. Corner solution - used to account for corner
4. Curvature solution - used to account for curvature discontinuity
5. Step solution - used to cancel step in  $\psi$  at corner

Homogeneous solutions. - The required solutions are axially symmetric solutions of the wave equation, homogeneous in the space variables. The order of homogeneity is integral (1 and 2) in the first two cases, and half-integral ( $1/2$ ,  $3/2$ ,  $-1/2$ ) in the others. Such solutions have been studied in detail by Hayes (reference 8). For present purposes  $\phi^{(m)}$ ,

the solution homogeneous of order  $m$ , can be obtained by taking the source distribution  $S(x)$  in equation (15) proportional to  $x^m$ . It is convenient to choose the source strength as

$$S^{(m)}(x) = \frac{C}{m!} x^m \quad (16)$$

where  $C$  is a normalization constant, so that solutions of various orders are related by

$$\phi^{(m-p)} = \left( \frac{\partial}{\partial x} \right)^p \phi^{(m)} \quad (17)$$

For integral  $m$ , the solutions have simplest form if the normalization constant  $C$  is taken to be unity. Then using various relations for the hypergeometric function (see, for example, reference 9) the solutions are found to be given by

$$\phi^{(m)}(x, r) = - \frac{x^m}{1 \cdot 3 \dots (2m-1)} (1-t^2)^{\frac{m+\frac{1}{2}}{2}} F \left( \frac{m+1}{2}, \frac{m+2}{2}; \frac{m+3}{2}; 1-t^2 \right) \quad (18)$$

Here the conical variable

$$t = \frac{\beta r}{x} \quad (19)$$

is the ratio of the tangent of the polar angle to the tangent of the Mach angle, and so varies from zero on the axis to unity at the Mach cone. For integral  $m$ , the hypergeometric functions which occur in equation (18) can be expressed in terms of products of  $\sqrt{1-t^2}$  and  $\operatorname{sech}^{-1} t$  with polynomials in  $t^2$ . The first two required basic solutions are obtained by setting  $m$  equal to 1 and 2, which gives:

#### Linear source solution ( $m = 1$ )

$$\begin{aligned} \phi &= -x (\operatorname{sech}^{-1} t - \sqrt{1-t^2}) & \varphi_{xx} &= -\frac{1}{x} \frac{1}{\sqrt{1-t^2}} \\ \varphi_x &= -\operatorname{sech}^{-1} t & \varphi_{xr} &= \frac{\beta}{x} \frac{1}{t \sqrt{1-t^2}} \\ \varphi_r &= \beta \frac{\sqrt{1-t^2}}{t} & \varphi_{rr} &= -\frac{\beta^2}{x} \frac{1}{t^2 \sqrt{1-t^2}} \end{aligned} \quad \left. \right\} \quad (20)$$

Quadratic source solution ( $m = 2$ )

$$\left. \begin{aligned} \Phi &= -\frac{1}{2}x^2 \left[ \left(1 + \frac{1}{2}t^2\right) \operatorname{sech}^{-1}t - \frac{3}{2}\sqrt{1-t^2} \right] & \Phi_{xx} &= -\operatorname{sech}^{-1}t \\ \Phi_x &= -x(\operatorname{sech}^{-1}t - \sqrt{1-t^2}) & \Phi_{xr} &= \beta \frac{\sqrt{1-t^2}}{t} \\ \Phi_r &= \frac{\beta}{2}x \left( \frac{\sqrt{1-t^2}}{t} - t \operatorname{sech}^{-1}t \right) & \Phi_{rr} &= -\frac{\beta^2}{2} \left( \frac{\sqrt{1-t^2}}{t^2} + \operatorname{sech}^{-1}t \right) \end{aligned} \right\} \quad (21)$$

For half-integral  $m$ , it is convenient to choose the normalization constant  $C$  as  $\sqrt{2/\pi}$ , so that the solutions have simple values at the Mach cone. (The difference in normalization for integral and half-integral  $m$  is of no concern, because the connection between them is never used.) Transforming the hypergeometric function into a more useful form for this case gives

$$\Phi^{(m)}(x, r) = -x^m \frac{\sqrt{2}(1-t)^{\frac{m+1}{2}}}{\Gamma\left(\frac{m+3}{2}\right) \sqrt{1+t}} F\left(\frac{1}{2}, m+1; m+\frac{3}{2}; \frac{1-t}{1+t}\right) \quad (22)$$

The hypergeometric functions occurring here can be expressed in terms of products of complete elliptic integrals and algebraic functions of  $t$ . The remaining three required basic solutions are obtained by setting  $m$  equal to  $1/2$ ,  $3/2$ , and  $-1/2$ . For convenience, asymptotic values valid just inside the Mach cone (where  $t = 1$ ) are also given below:

Corner solution ( $m = 1/2$ )

$$\left. \begin{aligned} \Phi &= -\sqrt{x} \frac{4\sqrt{2}}{\pi} \sqrt{1+t} (K-E) & \sim 0 \\ \Phi_x &= -\frac{1}{\sqrt{x}} \frac{2\sqrt{2}}{\pi} \frac{1}{\sqrt{1+t}} K & \sim -\frac{1}{\sqrt{x}} \\ \Phi_r &= \frac{\beta}{\sqrt{x}} \frac{2\sqrt{2}}{\pi} \frac{1}{\sqrt{1+t}} \left( \frac{1+t}{t} E - K \right) & \sim \frac{\beta}{\sqrt{x}} \\ \Phi_{xx} &= \frac{1}{x^{3/2}} \frac{\sqrt{2}}{\pi} \frac{1}{(1-t)\sqrt{1+t}} (K-E) & \sim \frac{1}{8} \frac{1}{x^{3/2}} \\ \Phi_{xr} &= \frac{\beta}{x^{3/2}} \frac{\sqrt{2}}{\pi} \frac{1}{(1-t)\sqrt{1+t}} \left( \frac{1}{t} E - K \right) & \sim \frac{3}{8} \frac{\beta}{x^{3/2}} \\ \Phi_{rr} &= -\frac{\beta^2}{x^{3/2}} \frac{\sqrt{2}}{\pi} \frac{1}{(1-t)\sqrt{1+t}} \left( \frac{2-t^2}{t^2} E - \frac{2-t}{t} K \right) & \sim -\frac{7}{8} \frac{\beta^2}{x^{3/2}} \end{aligned} \right\} \quad (23)$$

Curvature solution ( $m = 3/2$ )

$$\begin{aligned}
 \Phi &= -x^{3/2} \frac{8\sqrt{2}}{9\pi} \sqrt{1+t} [(3+t) K - 4E] & \sim 0 \\
 \Phi_x &= -\sqrt{x} \frac{4\sqrt{2}}{\pi} \sqrt{1+t} (K-E) & \sim 0 \\
 \Phi_r &= \beta \sqrt{x} \frac{4\sqrt{2}}{3\pi} \sqrt{1+t} \left( \frac{1}{t} E - K \right) & \sim 0 \\
 \Phi_{xx} &= -\frac{1}{\sqrt{x}} \frac{2\sqrt{2}}{\pi} \frac{1}{\sqrt{1+t}} K & \sim -\frac{1}{\sqrt{x}} \\
 \Phi_{xr} &= \frac{\beta}{\sqrt{x}} \frac{2\sqrt{2}}{\pi} \frac{1}{\sqrt{1+t}} \left( \frac{1+t}{t} E - K \right) & \sim \frac{\beta}{\sqrt{x}} \\
 \Phi_{rr} &= -\frac{\beta^2}{\sqrt{x}} \frac{2\sqrt{2}}{3\pi} \frac{1}{\sqrt{1+t}} \left( 2\frac{1+t}{t^2} E - \frac{2-t}{t} K \right) & \sim -\frac{\beta^2}{\sqrt{x}}
 \end{aligned}
 \quad \left. \right\} \quad (24)$$

Step solution ( $m = -1/2$ )

$$\begin{aligned}
 \Phi &= -\frac{1}{\sqrt{x}} \frac{2\sqrt{2}}{\pi} \frac{1}{\sqrt{1+t}} K & \sim -\frac{1}{\sqrt{x}} \\
 \Phi_x &= \frac{1}{x^{3/2}} \frac{\sqrt{2}}{\pi} \frac{1}{(1-t)\sqrt{1+t}} (K-E) & \sim \frac{1}{8} \frac{1}{x^{3/2}} \\
 \Phi_r &= \frac{\beta}{x^{3/2}} \frac{\sqrt{2}}{\pi} \frac{1}{(1-t)\sqrt{1+t}} \left( \frac{1}{t} E - K \right) & \sim \frac{3}{8} \frac{\beta}{x^{3/2}}
 \end{aligned}
 \quad \left. \right\} \quad (25)$$

Here  $K$  and  $E$  are the complete elliptic integrals of first and second kind with modulus  $k = \sqrt{(1-t)/(1+t)}$ . The second derivatives of the step solution are not required.

Use of relations among second derivatives. - All three second derivatives of the first-order potential are required in order to carry out the second-order solution. (See equations (5b).) Considerable labor can be avoided by calculating directly only one of them, say  $\Phi_{xx}$ . Then  $\Phi_{xr}$  and  $\Phi_{rr}$  can be obtained from the equation of motion and tangency condition. Thus the first-order equation of motion (3) gives immediately an expression for  $\Phi_{rr}$ :

$$\varphi_{rr} = \beta^2 \varphi_{xx} - \frac{\varphi_r}{r} \quad (26)$$

Differentiating the first-order tangency condition (equation (10)) with respect to  $x$  gives an expression for  $\varphi_{xr}$  on the surface of the body:

$$\varphi_{xr} = R'' - R' \varphi_{rr} \quad \text{at } r = R(x) \quad (27)$$

The computing forms described later incorporate this simplification.

Tables of basic solutions. - With this simplification, the five basic solutions and their required derivatives comprise 13 distinct functions. Each is a power of  $x$  multiplied by a function of  $t$  alone. Thus, associated with the linear and quadratic source solutions are the following six functions of  $t$ , which, as indicated, play different roles in the two solutions:

<u>Symbol</u>	<u>Functional form</u>	<u>Role in quadratic source solution</u>	<u>Role in linear source solution</u>	
$a(t)$	$\left(\frac{1}{2} + \frac{1}{4} t^2\right) \operatorname{sech}^{-1} t - \frac{3}{4} \sqrt{1-t^2}$	$-\varphi/x^2$	---	$\operatorname{Sech}^{-1} t = \ln\left(\frac{1}{t} + \sqrt{\frac{1}{t^2}-1}\right)$
$b(t)$	$\operatorname{sech}^{-1} t - \sqrt{1-t^2}$	$-\varphi_x/x$	$-\varphi/x$	
$c(t)$	$\frac{1}{2} \left( \frac{\sqrt{1-t^2}}{t} - t \operatorname{sech}^{-1} t \right)$	$\varphi_r/\beta x$	---	
$d(t)$	$\operatorname{sech}^{-1} t$	$-\varphi_{xx}$	$-\varphi_x$	
$e(t)$	$\frac{\sqrt{1-t^2}}{t}$	$(\varphi_{xr}/\beta)$	$\varphi_r/\beta$	
$f(t)$	$\frac{1}{\sqrt{1-t^2}}$	---	$-x\varphi_{xx}$	

These functions are tabulated in table I for  $t$  ranging from 0.100 to 0.940 by increments of 0.001.<sup>4</sup> Values are given to six significant figures or seven decimals, whichever is the lesser, and are believed to be correct to within one-half unit in the last place. Linear interpolation results in errors of no more than three units in the last place except near the beginning and end of the table.

<sup>4</sup>Tables I and II are modeled after unpublished tables for calculating first-order supersonic flow past inclined bodies which were prepared for the author at the Rand Corporation.

Likewise, associated with the corner, curvature, and step solutions are the following seven functions of  $t$ :

<u>Symbol</u>	<u>Functional form</u>	<u>Role in curvature solution</u>	<u>Role in corner solution</u>	<u>Role in step solution</u>		
$g(t)$	$\frac{8\sqrt{2}}{9\pi} \sqrt{1+t} [(3t+1) K - 4E]$	$-\varphi/x^{3/2}$	---	---	(29)	
$h(t)$	$\frac{4\sqrt{2}}{\pi} \sqrt{1+t} (K-E)$	$-\varphi_x/\sqrt{x}$	$-\varphi/\sqrt{x}$	---		
$i(t)$	$\frac{4\sqrt{2}}{3\pi} \sqrt{1+t} \left(\frac{1}{t} E - K\right)$	$\varphi_r/\beta\sqrt{x}$	---	---		
$j(t)$	$\frac{2\sqrt{2}}{\pi} \frac{1}{\sqrt{1+t}} K$	$-\sqrt{x}\varphi_{xx}$	$-\sqrt{x}\varphi_x$	$-\sqrt{x}\varphi$		
$k(t)$	$\frac{2\sqrt{2}}{\pi} \frac{1}{\sqrt{1+t}} \left(\frac{1+t}{t} E - K\right)$	$(\sqrt{x}\varphi_{xr}/\beta)$	$\sqrt{x}\varphi_r/\beta$	---		
$l(t)$	$\frac{\sqrt{2}}{\pi} \frac{1}{(1-t)\sqrt{1+t}} (K - E)$	---	$x^{3/2}\varphi_{xx}$	$x^{3/2}\varphi_x$		
$m(t)$	$\frac{\sqrt{2}}{\pi} \frac{1}{(1-t)\sqrt{1+t}} \left(\frac{1}{t} E - K\right)$	---	$(x^{3/2}\varphi_{xr}/\beta)$	$x^{3/2}\varphi_r/\beta$		

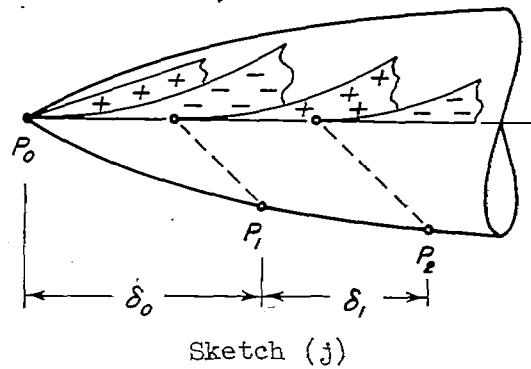
These functions are tabulated in table II for  $t$  ranging from 0.100 to 1.000 by increments of 0.001. The number of figures and accuracy are the same as for table I. Linear interpolation results in errors of no more than three units in the last place except for certain of the functions near the beginning of the table.

To facilitate interpolation, first forward differences are given without their algebraic sign in both tables. It should be noted that the differences are actually negative except in the case of the function  $f(t)$  in table I.

#### Choice of Intervals

The five basic solutions are superimposed to calculate the first-order potential  $\varphi$  and again to calculate the complementary function  $x$ .

The procedure, analogous to that of Kármán and Moore, is indicated in sketch (j) for a smooth pointed body. First, a linear source is added at the origin of strength sufficient to produce tangent flow just at the tip. Second, a quadratic source is added at the origin of strength (negative for a convex body), such that together with the linear source it produces tangent flow on the body at some distance  $\delta_0$  from the nose. Third, another quadratic source is added with its vertex shifted downstream so that its effect begins at the end of the first interval, and its strength is determined by imposing the tangency condition at some farther distance  $\delta_1$  along the body. Any corners or curvature discontinuities (or steps in the complementary function) must be accounted for by adding suitable strengths of the appropriate solutions, after which the superposition of quadratic sources continues as before.



Sketch (j)

The proper choice of intervals is of crucial importance. They should be taken as large as possible, because the computing labor increases nearly as the square of the number of intervals. On the other hand, the inaccuracy associated with using finite intervals rises with the square of their length, so that too large intervals lead to unacceptable error. It should be emphasized that the error considered here, which will be termed "numerical error," is the difference between the approximate second-order solution for finite intervals and the corresponding limiting solution for infinitesimal intervals; it is quite distinct from the difference between the second-order and exact solutions.

Fortunately, the tendency for numerical errors in successive intervals to accumulate is largely offset by the downstream damping of disturbances. Furthermore, successive numerical errors alternate in sign in most cases. Consequently, it has been found sufficient to formulate rules according to which each interval alone in an otherwise exact solution would cause no more than 1-percent numerical error. The entire second-order pressure distribution will then be determined correctly to within roughly 1 percent of the maximum pressure increment.

Simplification resulting from similarity.—The dependence of the first-order solution upon Mach number can be accounted for by the supersonic counterpart of the Göthert rule (reference 6), which is the similarity rule for linearized compressible flow. This similarity rule does not hold to second order. However, carrying out the usual similarity analysis shows that it holds approximately for the particular integral, which is the primary source of numerical error. (The similarity for the particular integral fails to be exact only to the extent to

which  $\beta$  differs from  $M$ , which is important only in the transonic range.) Therefore, any measure of numerical inaccuracy in the second-order solution may be expected to follow roughly the ordinary supersonic similarity rule. It is clear that this approximate result is adequate for estimating lengths of intervals, because moderate errors in interval length will not appreciably affect the solution. As a consequence, rules for choosing intervals which have been determined at one Mach number become universally valid if restated with the radius  $R$  replaced throughout by  $\beta R$ , the reduced radius of the supersonic similarity rule (or possibly  $MR$ , since the approximate similarity rule does not distinguish between  $\beta$  and  $M$ ). This conclusion, which greatly simplifies the formulation of rules, has been confirmed by a number of numerical calculations.

First interval for pointed body. - If a pointed body begins with a conical nose of finite length, the first interval is, of course, taken equal to the length of the cone. Otherwise, the meridian curve will ordinarily begin with finite curvature. For a specified limit of numerical error, the maximum permissible length of the first interval must be proportional to the initial radius of curvature, which is the primary length in the problem. The factor of proportionality will, of course, depend upon the shape of the body. If the meridian curve is analytic, dimensional analysis combined with the supersonic similarity rule indicates that the first interval is given by an expression of the form<sup>5</sup>

$$\delta_0 = \frac{1}{M|R_0''|} G_0 \left( \beta R_0', \frac{R_0'''}{\beta R_0''^2}, \dots \right) \quad (30)$$

Here  $R_0'$ ,  $R_0''$ ,  $R_0'''$  are the first three derivatives of  $R(x)$  evaluated at the vertex, and the dots indicate that no appreciable dependence upon higher derivatives is to be expected. Indeed, for slender smooth bodies even the second variable  $R_0'''/(\beta R_0''^2)$  is normally very small compared with the first. Hence it may be assumed that the function  $G_0$  does not depend significantly upon its second variable, so that the length of the first interval is given by

$$\delta_0 = \frac{1}{M|R_0''|} G_0(\beta R_0') \quad (31)$$

It is now clear that the body shape need not be analytic throughout the first interval; it is sufficient that no violent changes in curvature occur.

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<sup>5</sup>That the denominator should be taken as  $MR_0$  rather than  $\beta R_0$  is suggested by the result of equation (32).

The form of the function  $G_0$  can be determined by analysis, because the second-order solution at the end of the first interval of a general ogive can be calculated exactly as well as approximately if the interval is very short. Although the result is formidable, it simplifies greatly in the limiting case when  $\beta R_0'$  approaches unity (which corresponds physically to the Mach cone becoming tangent to the nose). In this case, for a relative numerical error  $\Delta\phi_x/\phi_x$  in streamwise velocity perturbation, the length of the first interval is

$$\delta_0 \sim \sqrt{\frac{40}{\gamma+1}} \frac{1}{M|R_0'''|} (1-\beta^2 R_0'^2) \sqrt{|\Delta\phi_x/\phi_x|} \quad \text{as } \beta R_0' \rightarrow 1 \quad (32)$$

Numerical examples show that this asymptotic form is, with a revised constant of proportionality, a good approximation to the function throughout the range of practical application. The relative numerical error at the end of the first interval will not exceed 1 percent if<sup>6</sup>

$$\delta_0 = \frac{1}{8} \frac{1}{M|R_0'''|} (1-\beta^2 R_0'^2)$$

(33)

It is conceivable that an unusual body shape might be encountered for which the curvature would change considerably over this length. If so, the above rule would not apply (the variable  $R_0'''/(\beta R_0'^2)$  in equation (30) would not be negligible), and some experimentation would be required to ascertain how much the interval should be reduced.

Internal intervals. - At any point on a smooth body, the length of the next interval will be proportional to the local radius, with the factor of proportionality depending upon the body shape in the vicinity of the point. If the meridian curve is analytic, dimensional analysis together with the supersonic similarity rule indicates that for a specified limit of numerical error the length of the interval from the nth to  $(n + 1)$ st point is given by

$$\delta_n = \beta R_n G_1(\beta R_n', \beta^2 R_n R_n'', \beta^3 R_n^2 R_n''', \dots) \quad (34)$$

The third variable here corresponds to the second variable in equation (30); its form is different because  $R$  rather than  $1/R''$  is taken as the primary reference length. (The second variable here has no counterpart in equation (30) because  $R$  is zero at the tip.) For a smooth slender body, the third variable is ordinarily very small, as

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<sup>6</sup>This rule ordinarily permits greater first intervals than the rule  $\delta_0 = 0.025/\beta$  times initial radius of curvature which was previously suggested in reference 4.

are all subsequent variables which involve higher derivatives. Then according to the argument used previously, the function  $G_1$  depends significantly upon only its first two variables. This conclusion is reinforced by the empirically determined fact that discontinuities in curvature must be accounted for separately, but not jumps in third and higher derivatives. Hence the  $n$ th interval is given by

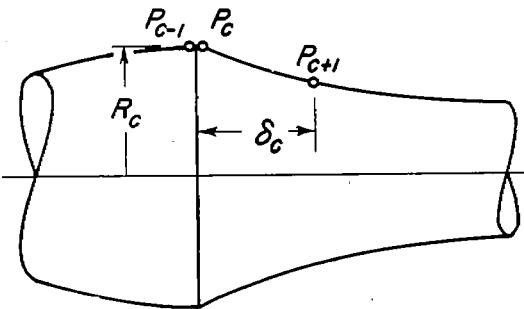
$$\delta_n = \beta R_n G_1(\beta R_n', \beta^2 R_n R_n'') \quad (35)$$

As before, the assumption that the body is analytic can now be replaced by the requirement that no violent changes in curvature occur.

Analytic determination of the function  $G_1$  seems impractical. Its detailed form could be determined experimentally by calculating a number of solutions using intervals of various lengths. However, experience suggests that for the body shapes encountered in practice  $G_1$  may be taken as a constant. The relative numerical error will apparently not exceed 1 percent if internal intervals for bodies without corners are chosen so that

$$\boxed{\delta_n = \beta R_n} \quad (36)$$

Modification for corner or curvature discontinuity.— Two points must be chosen at any discontinuity in slope or curvature, one just on each side, as indicated in sketch (k). A corner so strongly affects the subsequent flow field that it has been found necessary to reduce the next interval. The relative error will apparently not exceed 1 percent if the interval following a corner is taken to be

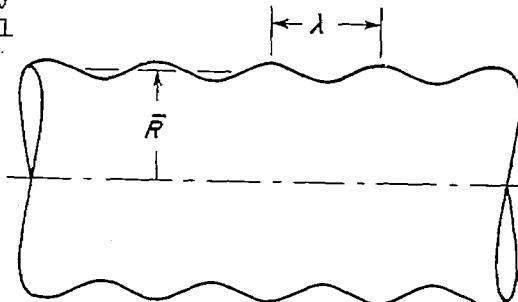


$$\delta_c = \frac{1}{2} \beta R_c \quad (37)$$

Sketch (k) where  $R_c$  is the radius at the corner. Thereafter, intervals can be chosen according to the rule for smooth bodies (equation (36)).

Limitations of rules.— These rules for choosing intervals are intended only as guides and must not be followed blindly. Although adequate for most bodies, they may fail for unusual shapes, particularly those having rapid changes in curvature. For example, the rule for choosing internal intervals (equation (36)) does not apply to the

corrugated body shown in sketch (l). In this case the variable  $\beta^3 R_n^2 R_n'''$  which was taken to be very small in equation (34) is proportional to  $(\bar{R}/\lambda)^2$ , and so becomes arbitrarily large as the corrugation wave length is reduced. It is clear physically that the interval should in this case be chosen as some fraction of the wave length. Fortunately, the fact that intervals have been taken too large usually reveals itself by excessive scatter in the final second-order results.



Sketch (l)

Also, the rules have been developed for the purpose of calculating flows at moderate or high supersonic speeds. They may accordingly become invalid at Mach numbers only slightly greater than unity, where they should involve the transonic similarity parameter,  $R'/\beta$ .

As in the case of solution by the method of characteristics, the only infallible rule (which may be invoked in case of doubt) is that the intervals are sufficiently small if further reduction causes no discernible change in the result.

The rules given above are believed to be somewhat conservative for normal shapes. In some cases, therefore, experience may indicate that the length of the intervals can be increased. It seems inadvisable, however, ever to double the prescribed values; not only is the scatter quadrupled, but successive errors then accumulate to such an extent that the result departs progressively farther from the true solution with distance downstream.

#### Description of Computing Forms

Standard computing forms have been prepared which largely reduce the second-order solution to routine calculation with a desk machine. Form A is used for bodies having continuous curvature. Form B is an insert to be pasted into form A to account for a corner or discontinuity in curvature. Provision is made for six points beyond the tip of a pointed body, which is adequate for most purposes. The forms can readily be extended to handle longer calculations.<sup>7</sup> Copies of the forms suitable for photosensitive reproduction are enclosed.

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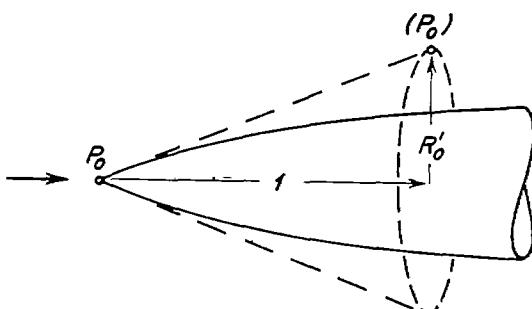
<sup>7</sup>Thus if one extra point is required, every row on each side of forms A and B which now extends to column P<sub>6</sub> (except rows (6m) to (6s), (6mm), and (6ss) of form A) is extended to form an additional column labeled P<sub>7</sub>, and below row (6w) of form A is inserted a new group of rows identical with rows (6a) to (6w) on the left and (6mm) to (6vw) on the right, but labeled (7-) and containing blanks only in column P<sub>7</sub>.

The desired values of Mach number and  $\gamma$  are entered at the top of form A, together with values of  $x, R, R'$ , and  $R''$  at points along the body chosen according to the rules formulated above. Then the form can be given to a computer together with tables I and II. The solution for a typical ogive or boattail can be calculated in from 5 to 10 hours.

As the solution progresses along the body, the results are found as differences of increasingly large numbers. Consequently, it is advisable to carry all computations to six significant figures or seven decimal places, whichever is the lesser. It is for this reason that tables I and II must be so extensive. It is not, of course, necessary to prescribe the problem with such accuracy; it is sufficient to give  $M, \gamma$ , and the body shape to three significant figures.

Details of form A. - The left half of form A is devoted chiefly to the calculation of the first-order potential  $\phi$  and its required derivatives. The particular integral  $\psi$  is also found in the last 23 rows of the left side. The right half gives a parallel calculation of the complementary function  $X$ . The second-order pressure coefficient is obtained in rows (63) to (73), and the corresponding first-order result, if required, in rows (74) to (83).

Following various preliminary calculations in rows (1) to (19), each group of from 10 to 13 rows bounded by double lines comprises a separate basic solution. The first such group (rows (1d) to (1w)) provides for a linear source solution beginning at the origin in case the body has a pointed tip. It may be noted that a stratagem has been introduced in calculating its effect at the tip. There both  $x$  and  $R$  are zero, so that the value of the conical variable  $t$  given by equation (19) would be indeterminate. This difficulty is surmounted by identifying values at the tip with those at the end of a tangent cone whose length is arbitrarily chosen as unity, as indicated in sketch (m). The requisite modification of given values in the first column is indicated by



Sketch (m)

strength of the solution (row (-s)) from the tangency condition; third, calculation of its contributions to  $-\phi$ ,  $-\phi_x$ ,  $\phi_r/\beta$ , and  $-\phi_{xx}$  (rows (-t) to (-w)) at each of the points  $P_0$  to  $P_6$ .

asterisks in rows (13), (14), and (16). Each of the subsequent six groups (coded (1-) to (6-)) provides a quadratic source solution, the first beginning at the origin. Each of these seven groups is separated into three subdivisions: first, determination of the conical variable  $t$  (row (-d)) and interpolation of the required functions from table I; second, calculation of the required

These separate contributions are added to obtain the corresponding complete first-order results in rows (20) to (23). Then equations (26) and (27) permit the calculation of the remaining two second derivatives,  $-\Phi_{rr}$  (row (27)) and  $\Phi_{xr}$  (row (29)). Finally, equations (5) for the particular integral are used to determine  $\psi_x/M^2$  (row (45)),  $\psi_r/M^2$  (row (49)), and  $-\psi/M^2$  (row (52)), the last being required only on each side of every corner.

On the right half, various quantities required in calculating the complementary function  $X$  are assembled in rows (53) to (60). There follow seven groups of three or four rows each which are the second-order counterparts of the adjacent first-order groups, a linear source solution in rows (0-) and quadratic source solutions in rows (1-) to (6-). For each group, the second-order tangency condition yields a weighting factor (row (-ss)) which multiplies the first-order results to give the corresponding contributions to the complementary function. Thus the contributions to  $-x_x$  and  $x_r/\beta$  are found in rows (-uu) and (-vv).<sup>8</sup> Adding these together with the components due to the particular integral gives the complete second-order velocity components  $-\phi_x$  (row (61)) and  $\phi_r/\beta$  (row (62)). Then the second-order pressure coefficient at each point is determined in row (73) from equation (14). The first-order pressure coefficient, if required, is likewise obtained in row (83).

Details of form B. - The left half of form B provides a corner solution (rows (C-)) followed by a curvature solution (rows (K-)) for the first-order potential. Both are inserted at a corner; only the latter is used at a curvature discontinuity. The two groups are similar in structure to those of form A, with the addition that  $\Phi_{xr}/\beta$  is also calculated (rows (-x)) for later use.

The right half of form B contains the corresponding corner and curvature solutions for the complementary function. In addition, a step solution is provided (rows (S-)) which, as discussed previously, is required in the complementary function to neutralize a step in the particular integral at a corner. This step solution is placed adjacent to the first-order corner solution with which it is associated. Similarly, the corner solution is placed adjacent to the first-order curvature solution, with which it is associated even if the body has no corner. The curvature solution is not required in the complementary function except at a corner. At a corner the curvature discontinuity is so great that it must be accounted for at least approximately in order to preserve numerical accuracy. Its strength cannot be calculated exactly in terms of previously determined quantities, but fortunately curvature and corner solutions are so intimately related that it

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<sup>8</sup>It may be noted that the coding is mnemonic to the extent that rows (-u) and (-v) are proportional to the first-order velocity perturbations in  $u$  and  $v$ , and rows (-uu) and (-vv) to the second-order values.

suffices to take them in the same ratio in the complementary function as in the first-order solution.

Use for first-order solution alone. - A very accurate first-order solution is found in the course of the second-order computation. The present scheme can therefore be simplified if only a first-order solution is desired. Except for rows (74) to (83), only the left halves of forms A and B are used, and form A can be terminated with row (22) and form B with row Cx (because curvature discontinuities need not be accounted for). Moreover, the following rows can be deleted from form A:

(7); (8); (16); all (-e)'s, (-h)'s, (-t)'s, and (-w)'s; and (20)

and the following from form B:

(Ce), and (Ct)

The restrictions on interval length can be considerably relaxed. An analysis similar to that described previously shows that the first interval for a pointed ogive can be taken as

$$\delta_0 = \frac{1}{3} \frac{R_0'}{|R_0''|} \sqrt{1 - \beta^2 R_0'^2} \quad (38)$$

A few numerical examples suggest that subsequent intervals can be taken at least twice as large as for a second-order solution, so that

$$\delta_n = \begin{cases} 2\beta R_n & \text{except just behind a corner} \\ \beta R_n & \text{just behind a corner} \end{cases} \quad (39)$$

#### PRACTICAL USE OF METHOD

The following instructions are intended to permit the reader to apply the method without reference to the preceding detailed discussion.

#### Applicability

The method gives both the first- and second-order velocities and pressures at the external surface of a body of revolution in supersonic flow provided that:

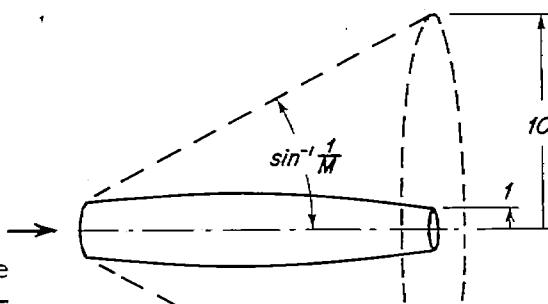
1. The body has a pointed nose, or has a sharp-edged open nose with purely supersonic external flow at the entrance, or is a boattail following an infinite cylinder.

2. The body contour is continuous (corners are permitted, but not steps), and has finite curvature (except at corners).

3. The slope of the contour is everywhere less than  $(M^2 - 1)^{-1/2}$ ,<sup>9</sup> the slope of the free-stream Mach cones.

In order to take advantage of the tables, the slope must in fact be nowhere greater than 94 percent of this value. Furthermore, the solution can be carried back only to the point at which the radius of the Mach cone from the nose has grown to ten times the local radius, as indicated in sketch (n) for an open-nosed body. The solution could be continued beyond this point only by extending the tables according to equations (28) and (29).

Choice of Points



Sketch (n)

For normal bodies, points on the body are chosen according to the following rules. These rules may fail if the curvature changes unusually rapidly; this will be revealed by excessive scatter in the second-order solution, which indicates that the intervals must be reduced.

1. Choose point  $P_0$  at the vertex of a pointed body.

2. If a pointed body has a conical nose of finite length, choose point  $P_1$  immediately behind the base of the cone. Otherwise, choose  $P_1$  at a distance behind the vertex no greater than

$$\delta_0 = \frac{1-\beta^2 R_0'^2}{8M |R_0''|}$$

where  $R_0'$  and  $R_0''$  are the slope and second derivative at the vertex.

3. Choose point  $P_1$  immediately behind the start of an open-nosed body or boattail.

---

<sup>9</sup>Although there is no absolute limitation on negative slope, the method becomes inaccurate when the magnitude of the maximum negative slope exceeds  $(M^2-1)^{1/2}$ .

4. Wherever the body has continuous curvature, choose point  $P_{n+1}$  beyond point  $P_n$  no farther than

$$\delta_n = \beta R_n$$

where  $R_n$  is the radius at  $P_n$ .

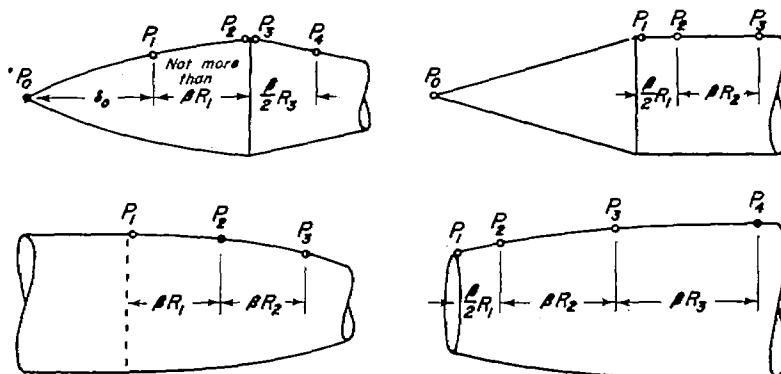
5. For a discontinuity in slope or curvature, reduce the preceding intervals if necessary so that a point falls exactly upon the discontinuity. Associate this point with the body shape just ahead of the discontinuity. Choose the next point at the same abscissa, but associate it with the body shape just behind the discontinuity. An exception arises, however, if the discontinuity follows a conical tip or infinitely long cylinder, or is the lip of an open-nosed body; then (as prescribed by rules 2 and 3) only a single point is required, and is associated with the body shape just behind the discontinuity.

6. Choose the first interval behind a corner no greater than

$$\delta_c = \frac{1}{2} \beta R_c$$

where  $R_c$  is the radius at the corner. A boattail or open-nosed body is to be regarded as starting with a corner if its initial slope is different from zero. The previous rules apply to subsequent intervals.

Examples of choice of points. - The choice of points for four typical bodies is indicated in sketch (o).



Sketch (o)

#### Preparation of Computing Form

Form A is prepared for computation in the following steps:

1. Enter the desired free-stream Mach number  $M$  in row (1) to three significant figures.

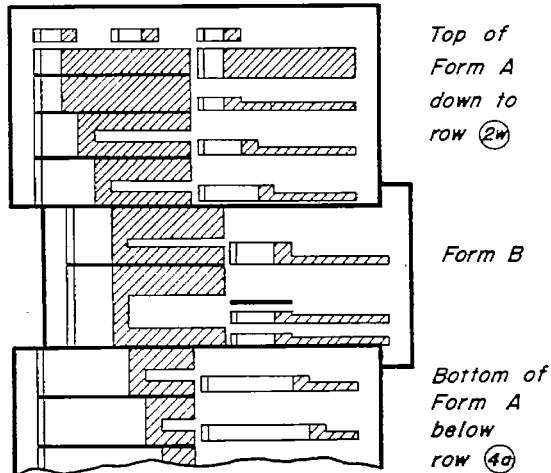
2. Enter the desired value of the adiabatic exponent  $\gamma$  in row (2) to three significant figures (1.40 for air).

3. In the column corresponding to each of the points  $P_n$  enter the abscissa in row (13), body radius in row (14), slope in row (15), and second derivative in row (16) to three significant figures.<sup>10</sup> However, in column  $P_0$  (which is used only for a pointed body) indeterminate forms are avoided by replacing the abscissa, radius, and second derivative by unity, the slope, and zero, respectively, as indicated on form A by asterisks. The origin for measuring abscissas must be taken at the tip of a pointed body, but is arbitrary for other shapes.<sup>11</sup> The unit of length is arbitrary, but it is usually convenient to measure in semicalibers.

4. If the body is not pointed, strike out column  $P_0$  and rows (Od) to (Ow) and (Oss) to (Ovv).

5. If point  $P_n$  lies just behind a corner or curvature discontinuity, cut out and discard all rows labeled (n-). Replace these by pasting in form B for a corner, or the portion of form B below the double line for a curvature discontinuity, with the first column alined below column  $P_n$  of form A. For example, sketch (p) shows schematically the modification required for a discontinuity between points  $P_2$  and  $P_3$ , as on the first body shown in sketch (o). Note that a boattail or open-nosed body is to be regarded as starting with a corner unless the initial slope is zero, and with a curvature discontinuity unless the initial curvature is zero.

#### Computing



Sketch (p)

The computing instructions on forms A and B are intended to be completely self-explanatory. As noted, all calculations should be carried to six significant figures or seven decimals, whichever

<sup>10</sup>Care should be taken to give  $R'$  and  $R''$  the proper algebraic sign.

<sup>11</sup>An exception arises in the unlikely case of an open-nosed body or boattail which starts with zero slope and curvature. In order to avoid indeterminate forms in this case, the origin must not coincide with the start of the contour.

is the lesser (regarding given data as exact to that accuracy). The tables should be interpolated linearly, noting that the first differences are given without algebraic sign.

Because the computations are rather involved, with only partial checks at rows (21) and (62), it has been found expedient when possible to have two computers carry out the same solution simultaneously with frequent comparisons. Typical shapes can be solved in from 5 to 10 hours.

### Results

The quantities of interest obtained at each point of the body are:

#### First-order quantities

$$\text{Row } (21) : -\Phi_x = 1 - \frac{u^{(1)}}{U}$$

$$\text{Row } (22) : \phi_r/\beta = \frac{1}{\beta} \frac{v^{(1)}}{U}$$

$$\text{Row } (83) : c_p^{(1)}$$

#### Second-order quantities

$$\text{Row } (62) : \phi_r/\beta = \frac{1}{\beta} \frac{v^{(2)}}{U}$$

$$\text{Row } (63) : 1 + \phi_x = \frac{u^{(2)}}{U}$$

$$\text{Row } (73) : c_p^{(2)}$$

Only three significant figures should be kept in the final results.

### Examples

Before calculating a new case, the reader may wish to check his computing procedure on the first few columns of a known solution. For this purpose, numerical values from various intermediate rows of the computing form are given below for a 6-caliber-long circular-arc ogive at a Mach number of 3. The significance of these rows is also indicated.

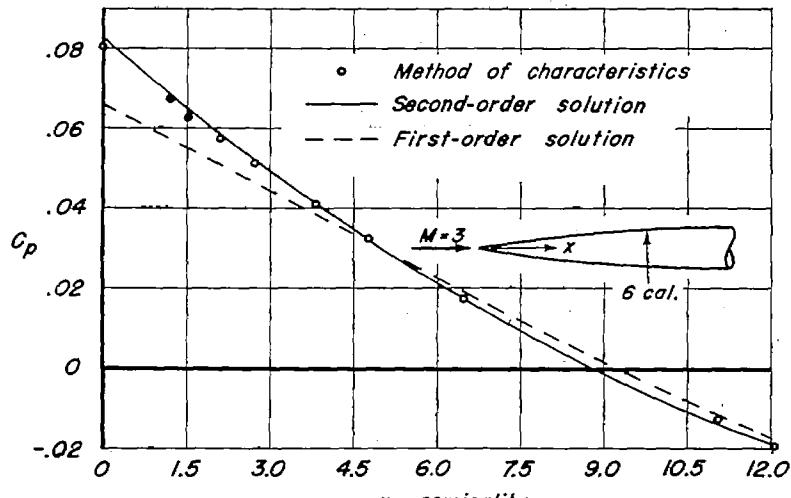
Dimensions are measured in semicalibers, and the intervals have been chosen slightly smaller than the limits prescribed by the rules in order to give simple values of  $x$ .

M:	1	3
$\gamma$ :	2	1.4

		$P_0$	$P_1$	$P_2$	$P_3$
$x:$	13	*1	2.00	2.80	3.90
R:	14	*.168	.307	.414	.546
$R'$ :	15	.168	.139	.128	.112
$R''$ :	16	*0	-.0142	-.0141	-.0141
$-\Phi:$	20	.0158906	.0305140	.0413536	.0549784
$-\Phi_x:$	21	.0441146	.0333807	.0295479	.0239671
$\Phi_r/\beta:$	22	.0593969	.0491439	.0452548	.0395979
$-\Phi_{xx}:$	23	.0364553	-.0001277	-.0011030	-.0052442
$-\psi_x/M^2:$	45	.0018064	-.0002293	-.0003804	-.0006239
$\psi_r/M^2:$	49	.0037346	-.0019991	-.0021893	-.0028234
$\rightarrow \phi_r/\beta:$	62	.0567766	.0475034	.0439176	.0386489
$\rightarrow 1+\phi_x:$	63	.950400	.963404	.968955	.975150
$c_p^{(2)}:$	73	.0830	.0606	.0506	.0403
$c_p^{(1)}:$	83	.0660	.0514	.0459	.0376

Note: The asterisks serve as a reminder that in column  $P_0$  the actual values of  $x, R$ , and  $R''$  must be replaced by 1, the value of  $R'$ , and 0, respectively.

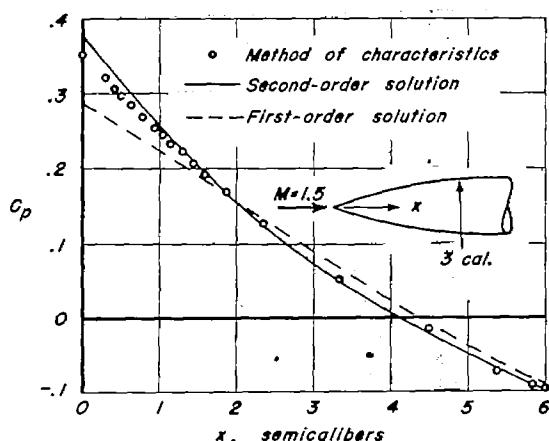
The first- and second-order pressure distributions for the complete ogive are shown in sketch (q) in comparison with a solution by the



Sketch (q)

numerical method of characteristics given by Rossow in reference 10.

As a further example, corresponding results are shown in sketch (r) for a 3-caliber ogive at a Mach number of 1.5.



Sketch (r)

Ames Aeronautical Laboratory  
 National Advisory Committee for Aeronautics  
 Moffett Field, Calif., May 12, 1952

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\* see Note p. 10. (3)

TABLE I.- LINEAR AND QUADRATIC SOURCE SOLUTIONS

t	a(t)	b(t)	c(t)	d(t)	e(t)	f(t)
.100	.757854	4800	1.99824	880	4.82528	5075
.101	.753054	4750	1.98834	881	4.77453	4878
.102	.748304	4700	1.97853	870	4.72475	4882
.103	.743604	4652	1.96883	881	4.67593	4781
.104	.738952	4604	1.95922	882	4.62602	4701
.105	.734348	4558	1.94970	842	4.58101	4814
.106	.729790	4512	1.94028	884	4.53487	4532
.107	.725278	4467	1.93094	885	4.48937	4448
.108	.720511	4424	1.92169	816	4.44509	4387
.109	.716387	4378	1.91253	888	4.40142	4281
.110	.712008	4338	1.90345	889	4.35851	4214
.111	.707670	4285	1.89446	882	4.31637	4141
.112	.703375	4235	1.88554	883	4.27496	4098
.113	.699120	4214	1.87671	875	4.23427	4000
.114	.694906	4175	1.86796	888	4.19427	3832
.115	.690731	4135	1.85928	880	4.15495	3883
.116	.686596	4088	1.85068	882	4.11630	3801
.117	.682498	4058	1.84216	846	4.07829	3788
.118	.678439	4023	1.83370	887	4.04091	3677
.119	.674416	3988	1.82533	891	4.00414	3617
.120	.670430	3950	1.81702	824	3.96797	5559
.121	.666480	3915	1.80878	817	3.93238	5502
.122	.662565	3880	1.80061	810	3.89736	5446
.123	.658685	3845	1.79251	804	3.86290	5382
.124	.654839	3812	1.78447	787	3.82898	5339
.125	.651027	3778	1.77650	790	3.79559	5288
.126	.647248	3746	1.76860	784	3.76271	5227
.127	.643502	3715	1.76076	778	3.73034	5188
.128	.639787	3683	1.75298	772	3.69845	5140
.129	.636104	3651	1.74526	765	3.66705	5083
.130	.632453	3621	1.73760	780	3.63612	5047
.131	.628832	3591	1.73000	783	3.60565	5002
.132	.625241	3560	1.72247	748	3.57563	2858
.133	.621681	3532	1.71499	743	3.54604	2815
.134	.618149	3502	1.70756	737	3.51689	2874
.135	.614647	3474	1.70019	781	3.48815	2882
.136	.611173	3446	1.69288	726	3.45983	2793
.137	.607727	3418	1.68562	720	3.43190	2753
.138	.604309	3381	1.67842	715	3.40437	2715
.139	.600918	3354	1.67127	710	3.37722	2677
.140	.597554	3327	1.66417	704	3.35045	2640
.141	.594217	3301	1.65713	700	3.32405	2604
.142	.590906	3285	1.65013	885	3.29801	2588
.143	.587621	3260	1.64318	889	3.27232	2535
.144	.584361	3244	1.63629	885	3.24697	2500
.145	.581127	3210	1.62944	880	3.22197	2468
.146	.577917	3185	1.62264	875	3.19129	2435
.147	.574732	3161	1.61589	871	3.17294	2403
.148	.571571	3137	1.60918	888	3.14891	2372
.149	.568434	3113	1.60252	881	3.12519	2341
.150	.565321	3090	1.59591	857	3.10178	2312
.151	.562231	3068	1.58934	853	3.07866	2282
.152	.559163	3044	1.58281	848	3.05584	2253
.153	.556119	3022	1.57633	843	3.03331	2223
.154	.553097	3000	1.56990	840	3.01106	2197
.155	.550097	2979	1.56350	835	2.98909	2170
.156	.547118	2956	1.55715	831	2.96739	2144
.157	.544162	2935	1.55084	827	2.94595	2117
.158	.541226	2914	1.54457	823	2.92478	2082
.159	.538312	2893	1.53834	819	2.90386	2058
.160	.535419		1.53215		2.88320	
					2.51927	
					6.16948	1.01305

TABLE I.—CONTINUED

t	a(t)	b(t)	c(t)	d(t)	e(t)	f(t)
.160	.535419 2673	1.53215 615	2.88320 2042	2.51927 632	6.16948 6835	1.01305 17
.161	.532546 2653	1.52600 611	2.86278 2017	2.51295 627	6.13015 3895	1.01322 17
.162	.529693 2633	1.51989 607	2.84261 1864	2.50668 624	6.09130 3838	1.01339 17
.163	.526860 2612	1.51382 604	2.82267 1870	2.50044 620	6.05292 3792	1.01356 17
.164	.524048 2724	1.50778 598	2.80297 1847	2.49424 616	6.01500 3746	1.01373 17
.165	.521254 2773	1.50179 586	2.78350 1824	2.48808 613	5.97754 3702	1.01390 17
.166	.518481 2755	1.49583 582	2.76126 1803	2.48195 608	5.94052 3658	1.01407 17
.167	.515726 2736	1.48991 588	2.74523 1880	2.47566 605	5.90393 3615	1.01424 18
.168	.512990 2717	1.48402 585	2.72643 1860	2.46981 602	5.86778 3573	1.01442 17
.169	.510273 2699	1.47817 581	2.70783 1838	2.46379 588	5.83205 3532	1.01459 18
.170	.507574 2680	1.47236 578	2.68945 1817	2.45780 585	5.79673 3481	1.01477 18
.171	.504894 2662	1.46658 575	2.67128 1798	2.45185 582	5.76182 3451	1.01495 18
.172	.502232 2645	1.46083 571	2.65330 1777	2.44593 589	5.72731 3412	1.01513 18
.173	.499587 2627	1.45512 567	2.63553 1758	2.44004 585	5.69319 3373	1.01531 18
.174	.496960 2609	1.44945 565	2.61795 1738	2.43419 582	5.65946 3335	1.01549 18
.175	.494351 2592	1.44380 561	2.60057 1719	2.42837 578	5.62611 3298	1.01567 19
.176	.491759 2575	1.43819 557	2.58338 1701	2.42258 575	5.59313 3262	1.01586 18
.177	.489184 2558	1.43262 555	2.56637 1683	2.41683 578	5.56051 3225	1.01604 18
.178	.486626 2541	1.42707 551	2.54954 1664	2.41110 583	5.52826 3180	1.01623 19
.179	.484083 2524	1.42156 548	2.53290 1647	2.40541 586	5.49636 3155	1.01642 18
.180	.481561 2508	1.41608 545	2.51643 1628	2.39975 583	5.46481 3120	1.01660 19
.181	.479052 2492	1.41063 542	2.50014 1612	2.39412 581	5.43361 3087	1.01679 20
.182	.476560 2476	1.40521 539	2.48402 1598	2.38851 557	5.40274 3054	1.01699 19
.183	.474084 2460	1.39982 535	2.46806 1578	2.38294 556	5.37220 3021	1.01718 19
.184	.471624 2444	1.39447 533	2.45228 1553	2.37739 551	5.34199 2888	1.01737 18
.185	.469180 2428	1.38914 530	2.43665 1540	2.37188 548	5.31210 2857	1.01756 20
.186	.466751 2414	1.38384 527	2.42119 1531	2.36639 546	5.28253 2827	1.01776 20
.187	.464337 2398	1.37857 525	2.40588 1515	2.36093 543	5.25326 2895	1.01796 19
.188	.461939 2383	1.37334 521	2.39073 1498	2.35550 540	5.22430 2865	1.01815 20
.189	.459556 2368	1.36813 518	2.37574 1485	2.35010 537	5.19565 2837	1.01835 20
.190	.457187 2353	1.36294 515	2.36089 1468	2.34473 535	5.16728 2807	1.01855 21
.191	.454834 2338	1.35779 512	2.34620 1455	2.33938 532	5.13921 2778	1.01876 20
.192	.452495 2324	1.35267 510	2.33165 1441	2.33406 529	5.11143 2750	1.01896 20
.193	.450171 2311	1.34757 507	2.31728 1428	2.32877 527	5.08393 2722	1.01916 21
.194	.447860 2295	1.34250 505	2.30298 1413	2.32350 524	5.05671 2693	1.01937 20
.195	.445565 2282	1.33745 501	2.28885 1398	2.31826 522	5.02976 2668	1.01957 21
.196	.443283 2268	1.33244 499	2.27486 1385	2.31304 518	5.00308 2641	1.01978 21
.197	.441015 2254	1.32745 496	2.26101 1372	2.30785 516	4.97667 2615	1.01999 21
.198	.438761 2241	1.32249 494	2.24729 1358	2.30269 514	4.95052 2590	1.02020 21
.199	.436520 2227	1.31755 491	2.23370 1345	2.29755 512	4.92462 2564	1.02041 21
.200	.434293 2213	1.31264 489	2.22025 1333	2.29243 506	4.89898 2539	1.02062 21
.201	.432080 2201	1.30775 486	2.20692 1321	2.28734 507	4.87359 2515	1.02083 22
.202	.429879 2187	1.30289 484	2.19371 1308	2.28227 504	4.84844 2490	1.02105 21
.203	.427692 2174	1.29805 481	2.18063 1298	2.27723 502	4.82354 2462	1.02126 22
.204	.425518 2161	1.29324 479	2.16767 1284	2.27221 499	4.79888 2443	1.02148 22
.205	.423357 2148	1.28845 476	2.15483 1271	2.26722 498	4.77445 2420	1.02170 22
.206	.421208 2135	1.28369 474	2.14212 1261	2.26224 495	4.75025 2397	1.02192 22
.207	.419072 2123	1.27895 471	2.12951 1248	2.25729 492	4.72626 2374	1.02214 22
.208	.416919 2111	1.27421 469	2.11703 1238	2.25237 491	4.70254 2352	1.02236 22
.209	.414838 2098	1.26955 467	2.10465 1228	2.24746 488	4.67902 2330	1.02258 23
.210	.412740 2085	1.26488 464	2.09239 1215	2.24258 486	4.65572 2308	1.02281 22
.211	.410634 2073	1.26024 462	2.08024 1204	2.23772 483	4.63263 2287	1.02303 23
.212	.408579 2062	1.25562 460	2.06820 1194	2.23289 482	4.60976 2266	1.02326 23
.213	.406517 2050	1.25102 458	2.05626 1183	2.22807 479	4.58710 2246	1.02349 23
.214	.404467 2039	1.24644 455	2.04443 1172	2.22328 478	4.56464 2225	1.02372 23
.215	.402428 2027	1.24189 453	2.03271 1163	2.21850 475	4.54239 2205	1.02395 23
.216	.400401 2015	1.23736 451	2.02108 1152	2.21375 473	4.52034 2185	1.02418 23
.217	.398386 2004	1.22285 449	2.00956 1142	2.20902 471	4.49849 2166	1.02441 23
.218	.396382 1992	1.22036 447	1.99814 1132	2.20431 468	4.47683 2146	1.02464 24
.219	.394390 1981	1.22389 444	1.98682 1122	2.19962 467	4.45537 2128	1.02488 24
.220	.392409	1.21945	1.97560	2.19495	4.43409	1.02512

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TABLE I.— CONTINUED

t	a(t)	b(t)	c(t)	d(t)	e(t)	f(t)
.220	.392409 1870	1.21945 442	1.97560 1113	2.19495 465	4.43409 2108	1.02512 25
.221	.390439 1859	1.21503 441	1.96447 1103	2.19030 463	4.41300 2090	1.02535 24
.222	.388480 1848	1.21062 468	1.95344 1084	2.18567 461	4.39210 2072	1.02559 24
.223	.386532 1837	1.20624 466	1.94250 1084	2.18106 458	4.37138 2053	1.02583 24
.224	.384595 1827	1.20168 434	1.93166 1078	2.17647 457	4.35085 2037	1.02607 23
.225	.382668 1815	1.19754 432	1.92090 1058	2.17190 455	4.33048 2018	1.02632 24
.226	.380753 1805	1.19322 430	1.91024 1058	2.16735 454	4.31030 2001	1.02656 24
.227	.378848 1895	1.18892 428	1.89966 1048	2.16281 451	4.29029 1885	1.02680 25
.228	.376953 1884	1.18464 426	1.88918 1041	2.15830 450	4.27044 1867	1.02705 25
.229	.375069 1873	1.18038 424	1.87877 1031	2.15380 447	4.25077 1851	1.02730 25
.230	.373196 1864	1.17614 422	1.86846 1023	2.14933 446	4.23126 1834	1.02755 25
.231	.371332 1853	1.17192 421	1.85823 1015	2.14487 444	4.21192 1815	1.02780 25
.232	.369479 1846	1.16771 418	1.84808 1008	2.14043 442	4.19274 1802	1.02805 25
.233	.367636 1833	1.16353 416	1.83802 998	2.13601 441	4.17372 1884	1.02830 25
.234	.365803 1820	1.15937 415	1.82803 990	2.13160 438	4.15466 1871	1.02856 25
.235	.363980 1813	1.15522 413	1.81813 982	2.12722 437	4.13615 1855	1.02881 25
.236	.362167 1803	1.15109 411	1.80830 974	2.12285 436	4.11760 1840	1.02907 25
.237	.360364 1794	1.14698 408	1.79856 967	2.11849 433	4.09920 1825	1.02933 25
.238	.358570 1784	1.14289 407	1.78889 959	2.11416 432	4.08095 1811	1.02958 27
.239	.356786 1775	1.13882 405	1.77930 952	2.10984 430	4.06284 1785	1.02985 26
.240	.355011 1765	1.13477 404	1.76978 944	2.10554 428	4.04489 1782	1.03011 26
.241	.353246 1756	1.13073 402	1.75034 937	2.10126 427	4.02707 1768	1.03037 26
.242	.351490 1745	1.12671 400	1.75097 930	2.09699 425	4.00941 1753	1.03063 27
.243	.349744 1737	1.12271 398	1.74167 922	2.09274 423	3.99188 1739	1.03090 27
.244	.348007 1728	1.11873 396	1.73245 916	2.08851 422	3.97449 1725	1.03117 26
.245	.346279 1718	1.11477 395	1.72329 908	2.08429 420	3.95724 1712	1.03143 27
.246	.344561 1710	1.11082 394	1.71421 901	2.08009 418	3.94012 1693	1.03170 28
.247	.342851 1701	1.10688 391	1.70520 895	2.07590 417	3.92314 1683	1.03198 27
.248	.341150 1692	1.10287 390	1.69625 888	2.07173 415	3.90629 1672	1.03225 27
.249	.339498 1683	1.09907 388	1.68737 881	2.06758 414	3.88957 1659	1.03252 28
.250	.337775 1674	1.09519 386	1.67856 874	2.06344 413	3.87298 1646	1.03280 27
.251	.336101 1665	1.09133 385	1.66982 868	2.05931 410	3.85652 1633	1.03307 28
.252	.334436 1657	1.08748 385	1.66114 862	2.05521 410	3.84019 1621	1.03335 28
.253	.332779 1648	1.08365 382	1.65252 855	2.05111 408	3.82398 1608	1.03363 28
.254	.331131 1640	1.07983 380	1.64397 849	2.04703 406	3.80789 1596	1.03391 28
.255	.329491 1631	1.07603 378	1.63548 842	2.04297 405	3.79193 1585	1.03419 28
.256	.327860 1623	1.07225 377	1.62706 837	2.03892 403	3.77608 1572	1.03447 29
.257	.326237 1615	1.06848 375	1.61869 830	2.03489 402	3.76036 1561	1.03476 28
.258	.324622 1606	1.06473 374	1.61039 824	2.03087 400	3.74475 1549	1.03504 29
.259	.323016 1598	1.06099 372	1.60215 818	2.02687 398	3.72926 1538	1.03533 29
.260	.321418 1580	1.05727 371	1.59397 813	2.02288 398	3.71388 1526	1.03562 29
.261	.319828 1582	1.05356 368	1.58584 808	2.01890 395	3.69862 1515	1.03591 29
.262	.318246 1574	1.04987 368	1.57778 801	2.01494 395	3.68347 1504	1.03620 29
.263	.316672 1565	1.04619 366	1.56977 795	2.01099 393	3.66843 1483	1.03649 29
.264	.315107 1558	1.04253 364	1.56182 790	2.00706 393	3.65350 1483	1.03678 29
.265	.313549 1550	1.03889 363	1.55392 784	2.00313 390	3.63867 1472	1.03708 29
.266	.311999 1542	1.03526 362	1.54608 778	1.99923 388	3.62396 1461	1.03737 29
.267	.310457 1535	1.03164 360	1.53830 773	1.99534 388	3.60935 1450	1.03767 29
.268	.308922 1527	1.02804 359	1.53057 768	1.99146 387	3.59485 1440	1.03797 29
.269	.307395 1519	1.02445 357	1.52289 762	1.98759 385	3.58045 1430	1.03827 29
.270	.305876 1511	1.02088 356	1.51527 757	1.98374 384	3.56615 1420	1.03857 29
.271	.304365 1504	1.01732 355	1.50770 752	1.97990 383	3.55195 1409	1.03888 29
.272	.302861 1497	1.01377 355	1.50018 746	1.97607 382	3.53786 1400	1.03918 29
.273	.301364 1488	1.01024 352	1.49272 742	1.97225 380	3.52386 1390	1.03949 29
.274	.299875 1481	1.00672 350	1.48530 738	1.96845 378	3.50996 1380	1.03979 29
.275	.298394 1475	1.00322 348	1.47794 731	1.96467 376	3.49616 1370	1.04010 29
.276	.296919 1468	.999731 3475	1.47063 727	1.96089 375	3.48246 1362	1.04041 29
.277	.295453 1460	.996256 3458	1.46336 722	1.95713 374	3.46884 1351	1.04072 29
.278	.293993 1453	.992794 3449	1.45614 718	1.95337 373	3.45533 1343	1.04104 29
.279	.292540 1445	.989345 3435	1.44898 712	1.94964 372	3.44190 1333	1.04135 29
.280	.291095	.985910	1.44186	1.94591	3.42857	1.04167

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TABLE I.—CONTINUED

t	a(t)	b(t)	c(t)	d(t)	e(t)	f(t)
.340	.215875	1084	.801291	2761	1.08689	483
.341	.214791	1080	.798530	2752	1.08196	480
.342	.213711	1075	.795778	2743	1.07706	477
.343	.212636	1069	.793035	2734	1.07219	474
.344	.211567	1065	.790301	2725	1.06735	472
.345	.210502	1060	.787576	2717	1.06253	470
.346	.209442	1055	.784899	2707	1.05774	477
.347	.208386	1050	.782152	2698	1.05297	474
.348	.207336	1046	.779454	2689	1.04823	472
.349	.206290	1041	.776764	2680	1.04351	469
.350	.205249	1037	.774084	2673	1.03882	467
.351	.204212	1032	.771411	2663	1.03415	464
.352	.203180	1027	.768748	2655	1.02951	462
.353	.202153	1023	.766093	2646	1.02489	458
.354	.201130	1018	.763447	2638	1.02030	457
.355	.200112	1013	.760809	2629	1.01573	454
.356	.199099	1008	.758180	2621	1.01119	453
.357	.198090	1004	.755559	2612	1.00666	449
.358	.197086	1000	.752947	2604	1.00217	448
.359	.196086	996	.750343	2596	.997692	4451
.360	.195090	991	.747747	2587	.993241	4428
.361	.194099	986	.745160	2579	.988813	4405
.362	.193113	983	.742581	2571	.984407	4382
.363	.192130	977	.740010	2563	.980025	4351
.364	.191153	974	.737447	2555	.975664	4328
.365	.190179	969	.734892	2548	.971326	4316
.366	.189210	965	.732346	2539	.967010	4294
.367	.188245	960	.729807	2531	.962716	4273
.368	.187285	957	.727276	2522	.958443	4250
.369	.186328	952	.724754	2515	.954193	4230
.370	.185376	948	.722239	2507	.949963	4208
.371	.184428	943	.719732	2499	.945755	4187
.372	.183485	940	.717233	2482	.941568	4165
.373	.182545	935	.714741	2463	.937402	4145
.374	.181610	931	.712258	2476	.933257	4125
.375	.180679	927	.709782	2459	.929132	4104
.376	.179752	923	.707313	2450	.925028	4084
.377	.178829	918	.704853	2452	.920944	4064
.378	.177910	915	.702400	2446	.916880	4044
.379	.176995	911	.699954	2438	.912836	4024
.380	.176081	907	.697516	2430	.908812	4004
.381	.175177	903	.695086	2423	.904808	3985
.382	.174274	900	.692663	2416	.900823	3965
.383	.173375	894	.690247	2408	.896858	3946
.384	.172481	891	.687839	2401	.892912	3927
.385	.171590	887	.685438	2393	.889895	3908
.386	.170703	883	.683045	2385	.885077	3888
.387	.169820	880	.680659	2379	.881188	3871
.388	.168940	875	.678280	2372	.877317	3852
.389	.168065	872	.675908	2365	.873465	3833
.390	.167193	867	.673543	2357	.869632	3815
.391	.166326	864	.671186	2351	.865817	3797
.392	.165462	860	.668835	2343	.862020	3778
.393	.164602	857	.666492	2335	.858241	3761
.394	.163745	852	.664156	2330	.854480	3743
.395	.162893	848	.661826	2322	.850737	3725
.396	.162044	845	.659504	2315	.847012	3708
.397	.161199	842	.657189	2309	.843304	3690
.398	.160357	838	.654880	2301	.839614	3674
.399	.159519	834	.652579	2295	.835940	3655
.400	.158685		.650284		.832284	
					1.56680	2.29129
						1.09109

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TABLE I.—CONTINUED

t	a(t)	b(t)	c(t)	d(t)	e(t)	f(t)
.400	.158685 <del>831</del>	.650284 <del>2288</del>	.832284 <del>8339</del>	1.56680 <del>273</del>	2.29129 <del>831</del>	1.09109 <del>52</del>
.401	.157854 <del>826</del>	.647996 <del>2281</del>	.828645 <del>8322</del>	1.56407 <del>272</del>	2.26448 <del>877</del>	1.09161 <del>52</del>
.402	.157028 <del>823</del>	.645715 <del>2274</del>	.825023 <del>8305</del>	1.56135 <del>271</del>	2.27771 <del>874</del>	1.09213 <del>53</del>
.403	.156205 <del>820</del>	.643441 <del>2268</del>	.821418 <del>8388</del>	1.55864 <del>271</del>	2.27097 <del>872</del>	1.09266 <del>52</del>
.404	.155385 <del>816</del>	.641173 <del>2261</del>	.817829 <del>8372</del>	1.55593 <del>270</del>	2.26425 <del>868</del>	1.09318 <del>53</del>
.405	.154569 <del>812</del>	.638912 <del>2254</del>	.814257 <del>8355</del>	1.55323 <del>270</del>	2.25757 <del>865</del>	1.09371 <del>53</del>
.406	.153757 <del>808</del>	.636698 <del>2248</del>	.810702 <del>8340</del>	1.55053 <del>269</del>	2.25092 <del>862</del>	1.09424 <del>54</del>
.407	.152948 <del>804</del>	.634410 <del>2241</del>	.807162 <del>8323</del>	1.54784 <del>268</del>	2.24430 <del>860</del>	1.09478 <del>53</del>
.408	.152142 <del>802</del>	.632169 <del>2234</del>	.803639 <del>8306</del>	1.54515 <del>265</del>	2.23770 <del>858</del>	1.09531 <del>54</del>
.409	.151340 <del>798</del>	.629935 <del>2228</del>	.800133 <del>8281</del>	1.54247 <del>265</del>	2.23114 <del>854</del>	1.09585 <del>54</del>
.410	.150542 <del>795</del>	.627707 <del>2221</del>	.796642 <del>8475</del>	1.53979 <del>267</del>	2.22460 <del>851</del>	1.09639 <del>54</del>
.411	.149747 <del>781</del>	.625486 <del>2215</del>	.793167 <del>8459</del>	1.53712 <del>267</del>	2.21809 <del>848</del>	1.09693 <del>54</del>
.412	.148956 <del>788</del>	.623271 <del>2208</del>	.789708 <del>8444</del>	1.53445 <del>265</del>	2.21161 <del>845</del>	1.09747 <del>55</del>
.413	.148168 <del>785</del>	.621062 <del>2201</del>	.786264 <del>8427</del>	1.53179 <del>265</del>	2.20516 <del>842</del>	1.09802 <del>55</del>
.414	.147383 <del>781</del>	.618861 <del>2195</del>	.782837 <del>8410</del>	1.52914 <del>265</del>	2.19874 <del>840</del>	1.09857 <del>55</del>
.415	.146602 <del>778</del>	.616665 <del>2189</del>	.779424 <del>8397</del>	1.52649 <del>265</del>	2.19234 <del>837</del>	1.09912 <del>55</del>
.416	.145824 <del>774</del>	.614476 <del>2183</del>	.776027 <del>8381</del>	1.52381 <del>264</del>	2.18597 <del>834</del>	1.09967 <del>55</del>
.417	.145050 <del>771</del>	.612293 <del>2176</del>	.772646 <del>8367</del>	1.52120 <del>264</del>	2.17963 <del>831</del>	1.10022 <del>56</del>
.418	.144279 <del>768</del>	.610117 <del>2171</del>	.769279 <del>8351</del>	1.51856 <del>263</del>	2.17332 <del>829</del>	1.10078 <del>56</del>
.419	.143511 <del>764</del>	.607946 <del>2164</del>	.765928 <del>8336</del>	1.51593 <del>262</del>	2.16703 <del>828</del>	1.10134 <del>56</del>
.420	.142747 <del>761</del>	.605782 <del>2157</del>	.762592 <del>8322</del>	1.51331 <del>262</del>	2.16077 <del>823</del>	1.10190 <del>56</del>
.421	.141986 <del>757</del>	.603625 <del>2152</del>	.759270 <del>8307</del>	1.51069 <del>262</del>	2.15454 <del>821</del>	1.10246 <del>57</del>
.422	.141229 <del>753</del>	.601473 <del>2145</del>	.755963 <del>8292</del>	1.50807 <del>261</del>	2.14833 <del>818</del>	1.10303 <del>56</del>
.423	.140474 <del>751</del>	.599328 <del>2139</del>	.752671 <del>8278</del>	1.50546 <del>261</del>	2.14215 <del>815</del>	1.10359 <del>57</del>
.424	.139723 <del>748</del>	.597189 <del>2133</del>	.749393 <del>8263</del>	1.50285 <del>260</del>	2.13600 <del>815</del>	1.10416 <del>58</del>
.425	.138975 <del>744</del>	.595056 <del>2127</del>	.746130 <del>8248</del>	1.50025 <del>260</del>	2.12987 <del>811</del>	1.10474 <del>57</del>
.426	.138231 <del>741</del>	.592929 <del>2120</del>	.742882 <del>8235</del>	1.49765 <del>259</del>	2.12376 <del>807</del>	1.10531 <del>58</del>
.427	.137490 <del>738</del>	.590809 <del>2115</del>	.739647 <del>8220</del>	1.49506 <del>259</del>	2.11769 <del>806</del>	1.10589 <del>57</del>
.428	.136752 <del>735</del>	.588694 <del>2108</del>	.736427 <del>8206</del>	1.49247 <del>258</del>	2.11163 <del>803</del>	1.10646 <del>58</del>
.429	.136017 <del>732</del>	.586585 <del>2102</del>	.733221 <del>8182</del>	1.48989 <del>258</del>	2.10560 <del>800</del>	1.10705 <del>58</del>
.430	.135285 <del>728</del>	.584483 <del>2097</del>	.730029 <del>8178</del>	1.48731 <del>257</del>	2.09960 <del>798</del>	1.10763 <del>58</del>
.431	.134537 <del>726</del>	.582386 <del>2091</del>	.726851 <del>8164</del>	1.48474 <del>257</del>	2.09362 <del>795</del>	1.10822 <del>58</del>
.432	.133831 <del>722</del>	.580295 <del>2084</del>	.723687 <del>8151</del>	1.48217 <del>257</del>	2.08767 <del>793</del>	1.10880 <del>58</del>
.433	.133109 <del>719</del>	.578211 <del>2078</del>	.720536 <del>8137</del>	1.47960 <del>256</del>	2.08174 <del>790</del>	1.10939 <del>58</del>
.434	.132390 <del>715</del>	.576132 <del>2073</del>	.717399 <del>8123</del>	1.47704 <del>255</del>	2.07584 <del>788</del>	1.10998 <del>60</del>
.435	.131675 <del>713</del>	.574059 <del>2067</del>	.714276 <del>8110</del>	1.47449 <del>255</del>	2.06995 <del>785</del>	1.11058 <del>60</del>
.436	.130942 <del>710</del>	.571992 <del>2061</del>	.711166 <del>8098</del>	1.47194 <del>255</del>	2.06410 <del>784</del>	1.11118 <del>60</del>
.437	.130252 <del>708</del>	.569931 <del>2055</del>	.708070 <del>8083</del>	1.46939 <del>254</del>	2.05826 <del>801</del>	1.11178 <del>60</del>
.438	.129516 <del>704</del>	.567876 <del>2050</del>	.704987 <del>8070</del>	1.46685 <del>254</del>	2.05245 <del>778</del>	1.11238 <del>60</del>
.439	.128842 <del>700</del>	.565826 <del>2044</del>	.701917 <del>8058</del>	1.46431 <del>253</del>	2.04667 <del>777</del>	1.11298 <del>61</del>
.440	.128142 <del>698</del>	.563782 <del>2038</del>	.698861 <del>8044</del>	1.46178 <del>253</del>	2.04090 <del>774</del>	1.11359 <del>61</del>
.441	.127444 <del>694</del>	.561744 <del>2032</del>	.695817 <del>8031</del>	1.45925 <del>252</del>	2.03516 <del>771</del>	1.11420 <del>61</del>
.442	.126750 <del>691</del>	.559712 <del>2027</del>	.692786 <del>8017</del>	1.45673 <del>252</del>	2.02915 <del>770</del>	1.11481 <del>61</del>
.443	.126059 <del>688</del>	.557685 <del>2021</del>	.689769 <del>8005</del>	1.45421 <del>252</del>	2.02375 <del>767</del>	1.11542 <del>62</del>
.444	.125371 <del>685</del>	.555664 <del>2015</del>	.686764 <del>8002</del>	1.45169 <del>251</del>	2.01808 <del>765</del>	1.11604 <del>62</del>
.445	.124685 <del>682</del>	.553649 <del>2008</del>	.683772 <del>7990</del>	1.44918 <del>251</del>	2.01243 <del>763</del>	1.11666 <del>62</del>
.446	.124003 <del>678</del>	.551640 <del>2005</del>	.680792 <del>7987</del>	1.44667 <del>250</del>	2.00600 <del>761</del>	1.11728 <del>62</del>
.447	.123324 <del>677</del>	.549635 <del>1998</del>	.677825 <del>7984</del>	1.44417 <del>249</del>	2.00119 <del>758</del>	1.11790 <del>63</del>
.448	.122647 <del>675</del>	.547637 <del>1993</del>	.674871 <del>7982</del>	1.44167 <del>249</del>	1.99561 <del>755</del>	1.11853 <del>62</del>
.449	.121974 <del>670</del>	.545644 <del>1987</del>	.671929 <del>7989</del>	1.43918 <del>249</del>	1.99005 <del>754</del>	1.11915 <del>63</del>
.450	.121304 <del>668</del>	.543657 <del>1982</del>	.669000 <del>79818</del>	1.43669 <del>248</del>	1.98451 <del>752</del>	1.11978 <del>64</del>
.451	.120636 <del>665</del>	.541675 <del>1978</del>	.666032 <del>79805</del>	1.43420 <del>248</del>	1.97899 <del>750</del>	1.12042 <del>63</del>
.452	.119971 <del>661</del>	.539699 <del>1971</del>	.663177 <del>79822</del>	1.43172 <del>248</del>	1.97349 <del>748</del>	1.12105 <del>64</del>
.453	.119310 <del>658</del>	.537728 <del>1965</del>	.660285 <del>79811</del>	1.42924 <del>248</del>	1.96801 <del>745</del>	1.12169 <del>64</del>
.454	.118651 <del>655</del>	.535763 <del>1960</del>	.657404 <del>79808</del>	1.42676 <del>247</del>	1.96256 <del>744</del>	1.12233 <del>65</del>
.455	.117995 <del>653</del>	.533803 <del>1954</del>	.654535 <del>79857</del>	1.42429 <del>246</del>	1.95712 <del>741</del>	1.12298 <del>64</del>
.456	.117342 <del>650</del>	.531849 <del>1948</del>	.651678 <del>79845</del>	1.42163 <del>246</del>	1.95171 <del>738</del>	1.12362 <del>65</del>
.457	.116692 <del>648</del>	.529900 <del>1944</del>	.648833 <del>79833</del>	1.41937 <del>246</del>	1.94632 <del>738</del>	1.12427 <del>65</del>
.458	.116044 <del>645</del>	.527956 <del>1938</del>	.646000 <del>79822</del>	1.41691 <del>246</del>	1.94094 <del>735</del>	1.12492 <del>65</del>
.459	.115399 <del>641</del>	.526018 <del>1933</del>	.643178 <del>79810</del>	1.41445 <del>245</del>	1.93599 <del>733</del>	1.12557 <del>66</del>
.460	.114758	.524085	.640368	1.41200	1.93026	1.12623

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TABLE I.—CONTINUED

t	a(t)	b(t)	c(t)	d(t)	e(t)	f(t)
.460	.114758 <b>533</b>	.524085 <b>1828</b>	.640368 <b>2784</b>	1.41200 <b>244</b>	1.93026 <b>531</b>	1.12623 <b>66</b>
.461	.114119 <b>533</b>	.522157 <b>1822</b>	.637570 <b>2787</b>	1.40956 <b>245</b>	1.92495 <b>530</b>	1.12689 <b>66</b>
.462	.113483 <b>534</b>	.520235 <b>1817</b>	.634783 <b>2775</b>	1.40711 <b>246</b>	1.91965 <b>527</b>	1.12755 <b>66</b>
.463	.112849 <b>535</b>	.518318 <b>1812</b>	.632008 <b>2764</b>	1.40468 <b>244</b>	1.91438 <b>525</b>	1.12821 <b>67</b>
.464	.112219 <b>535</b>	.516406 <b>1806</b>	.629244 <b>2753</b>	1.40224 <b>243</b>	1.90913 <b>524</b>	1.12888 <b>67</b>
.465	.111591 <b>525</b>	.514500 <b>1801</b>	.626491 <b>2741</b>	1.39981 <b>245</b>	1.90389 <b>521</b>	1.12955 <b>67</b>
.466	.110966 <b>525</b>	.512599 <b>1807</b>	.623750 <b>2731</b>	1.39738 <b>242</b>	1.89868 <b>520</b>	1.13022 <b>67</b>
.467	.110343 <b>519</b>	.510702 <b>1801</b>	.621019 <b>2719</b>	1.39496 <b>242</b>	1.89348 <b>517</b>	1.13089 <b>68</b>
.468	.109724 <b>517</b>	.508811 <b>1805</b>	.618300 <b>2708</b>	1.39254 <b>242</b>	1.88831 <b>516</b>	1.13157 <b>68</b>
.469	.109107 <b>515</b>	.506926 <b>1801</b>	.615592 <b>2697</b>	1.39012 <b>241</b>	1.88315 <b>514</b>	1.13225 <b>68</b>
.470	.108492 <b>511</b>	.505045 <b>1875</b>	.612895 <b>2687</b>	1.38771 <b>241</b>	1.87801 <b>512</b>	1.13293 <b>68</b>
.471	.107881 <b>509</b>	.503170 <b>1871</b>	.610208 <b>2675</b>	1.38530 <b>240</b>	1.87289 <b>510</b>	1.13362 <b>68</b>
.472	.107272 <b>506</b>	.501299 <b>1865</b>	.607533 <b>2665</b>	1.38290 <b>240</b>	1.86779 <b>508</b>	1.13430 <b>69</b>
.473	.106666 <b>504</b>	.499434 <b>1860</b>	.604868 <b>2654</b>	1.38090 <b>240</b>	1.86271 <b>506</b>	1.13499 <b>70</b>
.474	.106062 <b>501</b>	.497574 <b>1855</b>	.602214 <b>2643</b>	1.37810 <b>240</b>	1.85765 <b>505</b>	1.13569 <b>69</b>
.475	.105461 <b>503</b>	.495719 <b>1850</b>	.599571 <b>2635</b>	1.37570 <b>238</b>	1.85260 <b>503</b>	1.13638 <b>70</b>
.476	.104863 <b>503</b>	.493869 <b>1845</b>	.596938 <b>2622</b>	1.37331 <b>238</b>	1.84751 <b>501</b>	1.13708 <b>70</b>
.477	.104267 <b>503</b>	.492024 <b>1840</b>	.594316 <b>2611</b>	1.37093 <b>238</b>	1.84256 <b>499</b>	1.13778 <b>70</b>
.478	.103674 <b>500</b>	.490184 <b>1835</b>	.591705 <b>2602</b>	1.36854 <b>238</b>	1.83757 <b>497</b>	1.13848 <b>71</b>
.479	.103081 <b>500</b>	.488349 <b>1831</b>	.589103 <b>2591</b>	1.36616 <b>237</b>	1.83260 <b>495</b>	1.13919 <b>71</b>
.480	.102496 <b>503</b>	.486518 <b>1825</b>	.586512 <b>2580</b>	1.36379 <b>238</b>	1.82764 <b>494</b>	1.13990 <b>71</b>
.481	.101911 <b>503</b>	.484693 <b>1820</b>	.583932 <b>2570</b>	1.36141 <b>237</b>	1.82270 <b>492</b>	1.14061 <b>72</b>
.482	.101328 <b>500</b>	.482873 <b>1815</b>	.581362 <b>2560</b>	1.35904 <b>236</b>	1.81778 <b>490</b>	1.14133 <b>72</b>
.483	.100748 <b>507</b>	.481058 <b>1811</b>	.578802 <b>2550</b>	1.35668 <b>236</b>	1.81288 <b>488</b>	1.14205 <b>72</b>
.484	.100171 <b>505</b>	.479247 <b>1805</b>	.576252 <b>2540</b>	1.35432 <b>236</b>	1.80799 <b>487</b>	1.14277 <b>72</b>
.485	.0995958 <b>5724</b>	.477442 <b>1801</b>	.573712 <b>2530</b>	1.35196 <b>236</b>	1.80312 <b>485</b>	1.14349 <b>73</b>
.486	.0990234 <b>5700</b>	.475642 <b>1786</b>	.571182 <b>2520</b>	1.34960 <b>235</b>	1.79827 <b>484</b>	1.14422 <b>73</b>
.487	.0984534 <b>5674</b>	.473845 <b>1791</b>	.568662 <b>2510</b>	1.34725 <b>235</b>	1.79343 <b>482</b>	1.14495 <b>73</b>
.488	.0978660 <b>5649</b>	.472054 <b>1786</b>	.566152 <b>2500</b>	1.34490 <b>235</b>	1.78861 <b>480</b>	1.14568 <b>74</b>
.489	.0973211 <b>5624</b>	.470268 <b>1781</b>	.563652 <b>2490</b>	1.34255 <b>234</b>	1.78381 <b>478</b>	1.14642 <b>73</b>
.490	.0967587 <b>5593</b>	.468487 <b>1777</b>	.561162 <b>2481</b>	1.34021 <b>234</b>	1.77903 <b>477</b>	1.14715 <b>73</b>
.491	.0961988 <b>5575</b>	.466710 <b>1772</b>	.558681 <b>2471</b>	1.33787 <b>234</b>	1.77426 <b>476</b>	1.14790 <b>74</b>
.492	.0956443 <b>5549</b>	.464938 <b>1767</b>	.556210 <b>2461</b>	1.33553 <b>233</b>	1.76950 <b>473</b>	1.14864 <b>75</b>
.493	.0950864 <b>5523</b>	.463171 <b>1762</b>	.553749 <b>2452</b>	1.33320 <b>233</b>	1.76477 <b>473</b>	1.14939 <b>75</b>
.494	.0945339 <b>5501</b>	.461409 <b>1758</b>	.551297 <b>2442</b>	1.33087 <b>233</b>	1.76004 <b>470</b>	1.15014 <b>73</b>
.495	.0939838 <b>5477</b>	.459651 <b>1758</b>	.548855 <b>2432</b>	1.32854 <b>232</b>	1.75534 <b>469</b>	1.15089 <b>73</b>
.496	.0934361 <b>5452</b>	.457898 <b>1748</b>	.546423 <b>2424</b>	1.32622 <b>232</b>	1.75065 <b>467</b>	1.15164 <b>73</b>
.497	.0928909 <b>5428</b>	.456150 <b>1744</b>	.543999 <b>2415</b>	1.32390 <b>232</b>	1.74598 <b>466</b>	1.15240 <b>77</b>
.498	.0923481 <b>5403</b>	.454406 <b>1739</b>	.541586 <b>2405</b>	1.32158 <b>231</b>	1.74132 <b>464</b>	1.15317 <b>76</b>
.499	.0918078 <b>5380</b>	.452667 <b>1734</b>	.539181 <b>2395</b>	1.31927 <b>231</b>	1.73668 <b>463</b>	1.15393 <b>77</b>
.500	.0912698 <b>5355</b>	.450933 <b>1730</b>	.536786 <b>2385</b>	1.31696 <b>231</b>	1.73205 <b>461</b>	1.15470 <b>77</b>
.501	.0907342 <b>5332</b>	.449203 <b>1725</b>	.534403 <b>2377</b>	1.31465 <b>230</b>	1.72744 <b>460</b>	1.15547 <b>78</b>
.502	.0902010 <b>5308</b>	.447478 <b>1721</b>	.532023 <b>2367</b>	1.31235 <b>231</b>	1.72284 <b>458</b>	1.15625 <b>77</b>
.503	.0896701 <b>5284</b>	.445757 <b>1716</b>	.529656 <b>2359</b>	1.31004 <b>230</b>	1.71826 <b>456</b>	1.15702 <b>78</b>
.504	.0891417 <b>5262</b>	.444041 <b>1711</b>	.527297 <b>2349</b>	1.30774 <b>229</b>	1.71370 <b>455</b>	1.15780 <b>78</b>
.505	.0886155 <b>5238</b>	.442330 <b>1707</b>	.524948 <b>2341</b>	1.30545 <b>228</b>	1.70915 <b>454</b>	1.15859 <b>78</b>
.506	.0880927 <b>5214</b>	.440623 <b>1703</b>	.522607 <b>2331</b>	1.30316 <b>228</b>	1.70461 <b>452</b>	1.15938 <b>78</b>
.507	.0875703 <b>5191</b>	.438920 <b>1697</b>	.520276 <b>2285</b>	1.30087 <b>228</b>	1.70009 <b>451</b>	1.16016 <b>80</b>
.508	.0870512 <b>5168</b>	.437223 <b>1684</b>	.517953 <b>2314</b>	1.29858 <b>228</b>	1.69558 <b>449</b>	1.16096 <b>80</b>
.509	.0865344 <b>5145</b>	.435529 <b>1680</b>	.515639 <b>2305</b>	1.29630 <b>228</b>	1.69109 <b>447</b>	1.16176 <b>78</b>
.510	.0860199 <b>5122</b>	.433840 <b>1684</b>	.513334 <b>2296</b>	1.29401 <b>227</b>	1.68662 <b>447</b>	1.16255 <b>81</b>
.511	.0855077 <b>5099</b>	.432156 <b>1680</b>	.511038 <b>2287</b>	1.29174 <b>228</b>	1.68215 <b>444</b>	1.16336 <b>80</b>
.512	.0849978 <b>5075</b>	.430476 <b>1675</b>	.508751 <b>2279</b>	1.28946 <b>227</b>	1.67771 <b>444</b>	1.16416 <b>81</b>
.513	.0844902 <b>5053</b>	.428801 <b>1671</b>	.506472 <b>2270</b>	1.28719 <b>227</b>	1.67327 <b>442</b>	1.16497 <b>82</b>
.514	.0839849 <b>5031</b>	.427130 <b>1667</b>	.504202 <b>2262</b>	1.28492 <b>227</b>	1.66885 <b>440</b>	1.16579 <b>81</b>
.515	.0834818 <b>5008</b>	.425463 <b>1662</b>	.501940 <b>2255</b>	1.28265 <b>226</b>	1.66445 <b>438</b>	1.16660 <b>82</b>
.516	.0829810 <b>4986</b>	.423801 <b>1658</b>	.499687 <b>2244</b>	1.28039 <b>226</b>	1.66006 <b>438</b>	1.16742 <b>82</b>
.517	.0824824 <b>4963</b>	.422143 <b>1654</b>	.497443 <b>2235</b>	1.27813 <b>225</b>	1.65568 <b>437</b>	1.16824 <b>83</b>
.518	.0819861 <b>4941</b>	.420489 <b>1649</b>	.495207 <b>2228</b>	1.27587 <b>225</b>	1.65131 <b>435</b>	1.16907 <b>83</b>
.519	.0814920 <b>4918</b>	.418840 <b>1645</b>	.492979 <b>2210</b>	1.27361 <b>225</b>	1.64696 <b>433</b>	1.16990 <b>83</b>
.520	.0810001	.417195	.490760	1.27136	1.64263	1.17073

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TABLE I.- CONTINUED

t	a(t)	b(t)	c(t)	d(t)	e(t)	f(t)	
.520	.0810001	4896	.417195 1640	.490760 2211	1.27136 225	1.64263 435	1.17073 84
.521	.0805105	4875	.415555 1635	.488349 2205	1.26911 225	1.63830 430	1.17157 84
.522	.0800230	4852	.413919 1632	.486546 2184	1.26586 224	1.63400 430	1.17241 84
.523	.0795378	4830	.412287 1628	.484152 2177	1.26462 224	1.62970 428	1.17325 85
.524	.0790548	4809	.410659 1625	.481965 2178	1.26238 224	1.62542 427	1.17410 85
.525	.0785739	4787	.409036 1619	.479787 2170	1.26014 224	1.62115 425	1.17495 85
.526	.0780952	4766	.407417 1615	.477617 2162	1.25790 223	1.61689 424	1.17580 86
.527	.0776186	4745	.405802 1610	.475455 2153	1.25567 223	1.61265 423	1.17666 86
.528	.0771443	4723	.404192 1607	.473302 2146	1.25344 223	1.60842 422	1.17752 86
.529	.0766720	4701	.402585 1602	.471156 2138	1.25121 223	1.60420 420	1.17838 87
.530	.0762019	4679	.400983 1598	.469018 2130	1.24898 222	1.60000 420	1.17925 87
.531	.0757340	4658	.399385 1593	.466888 2122	1.24676 222	1.59580 418	1.18012 87
.532	.0752662	4635	.397792 1590	.464766 2114	1.24454 222	1.59162 416	1.18099 88
.533	.0748044	4613	.396202 1585	.462652 2107	1.24232 222	1.58746 416	1.18187 88
.534	.0743429	4595	.394617 1581	.460545 2098	1.24010 221	1.58330 414	1.18275 88
.535	.0738834	4574	.393036 1578	.458446 2080	1.23789 221	1.57916 413	1.18364 88
.536	.0734260	4554	.391458 1573	.456356 2084	1.23568 221	1.57503 411	1.18453 89
.537	.0729706	4532	.389885 1568	.454272 2075	1.23347 221	1.57092 411	1.18542 89
.538	.0725174	4512	.388317 1565	.452197 2068	1.23126 220	1.56681 408	1.18632 89
.539	.0720662	4490	.386752 1561	.450129 2061	1.22906 220	1.56272 408	1.18722 89
.540	.0716172	4471	.385191 1557	.448068 2053	1.22686 220	1.55864 407	1.18812 91
.541	.0711701	4450	.383634 1552	.446015 2045	1.22466 220	1.55457 406	1.18903 91
.542	.0707251	4428	.382082 1548	.443970 2038	1.22246 219	1.55051 404	1.18904 92
.543	.0702822	4410	.380534 1545	.441932 2030	1.22027 219	1.54647 403	1.19086 91
.544	.0698412	4388	.378989 1540	.439902 2023	1.21807 219	1.54244 403	1.19177 93
.545	.0694024	4368	.377449 1537	.437879 2016	1.21588 218	1.53841 401	1.19270 92
.546	.0689655	4348	.375912 1532	.435863 2008	1.21370 218	1.53440 398	1.19362 93
.547	.0685306	4328	.374380 1529	.433855 2001	1.21151 218	1.53041 398	1.19455 94
.548	.0680978	4309	.372851 1524	.431854 1884	1.20933 218	1.52642 398	1.19549 94
.549	.0676669	4288	.371327 1521	.429860 1887	1.20715 218	1.52244 398	1.19643 94
.550	.0672381	4268	.369806 1516	.427873 1878	1.20497 217	1.51848 395	1.19737 94
.551	.0668112	4248	.368290 1513	.425894 1872	1.20280 218	1.51453 394	1.19831 95
.552	.0663863	4220	.366777 1508	.423922 1865	1.20062 217	1.51059 393	1.19926 95
.553	.0659633	4200	.365259 1505	.421957 1858	1.19845 217	1.50666 392	1.20022 96
.554	.0655424	4181	.363764 1501	.419999 1851	1.19628 217	1.50274 391	1.20118 96
.555	.0651233	4171	.362263 1497	.418048 1844	1.19411 218	1.49883 390	1.20214 96
.556	.0647062	4151	.360766 1492	.416104 1838	1.19195 218	1.49493 389	1.20310 98
.557	.0643911	4132	.359274 1489	.414168 1830	1.18979 217	1.49104 387	1.20408 97
.558	.0638779	4113	.357784 1485	.412238 1823	1.18762 215	1.48717 387	1.20505 98
.559	.0634666	4093	.356289 1481	.410315 1816	1.18547 215	1.48330 388	1.20603 98
.560	.0630573	4075	.354818 1478	.408399 1810	1.18331 215	1.47945 384	1.20701 99
.561	.0626498	4054	.353340 1474	.406489 1802	1.18116 216	1.47561 384	1.20800 99
.562	.0622144	4037	.351866 1468	.404587 1805	1.17900 215	1.47177 382	1.20899 99
.563	.0618407	4018	.350397 1465	.402692 1800	1.17685 215	1.46795 381	1.20998 100
.564	.0614389	3998	.348931 1463	.400803 1804	1.17470 214	1.46414 380	1.21098 101
.565	.0610390	3978	.347468 1458	.398921 1875	1.17256 214	1.46034 379	1.21199 101
.566	.0606411	3951	.346010 1455	.397046 1868	1.17042 215	1.45655 378	1.21300 101
.567	.0602450	3924	.344555 1451	.395177 1862	1.16827 214	1.45276 377	1.21401 101
.568	.0598507	3904	.343104 1447	.393317 1855	1.16613 214	1.44899 376	1.21502 102
.569	.0594583	3885	.341657 1443	.391460 1848	1.16399 213	1.44523 375	1.21604 103
.570	.0590478	3867	.340214 1440	.389611 1842	1.16186 214	1.44148 374	1.21707 103
.571	.0586791	3848	.338774 1436	.387769 1835	1.15972 213	1.43774 373	1.21810 104
.572	.0582922	3828	.337338 1432	.385934 1830	1.15759 213	1.43401 372	1.21914 103
.573	.0579072	3802	.335906 1428	.384104 1822	1.15546 213	1.43029 371	1.22017 105
.574	.0575240	3813	.334478 1425	.382282 1816	1.15333 212	1.42698 370	1.22122 104
.575	.0571427	3786	.333053 1421	.380466 1810	1.15121 213	1.42288 370	1.22226 104
.576	.0567631	3777	.331632 1417	.378656 1803	1.14908 212	1.41918 368	1.22332 105
.577	.0563854	3760	.330215 1414	.376853 1797	1.14696 212	1.41550 367	1.22437 106
.578	.0560094	3742	.328801 1410	.375056 1791	1.14484 212	1.41183 367	1.22543 107
.579	.0556352	3723	.327391 1406	.373265 1784	1.14272 212	1.40816 366	1.22650 107
.580	.0552629		.325985	.371481	1.14060	1.40451	1.22757

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TABLE I.— CONTINUED

t	a(t)	b(t)	c(t)	d(t)	e(t)	f(t)
.640	.0359669	2743	.247973	1188	.275062	1444
.641	.0356926	2729	.246774	1186	.273618	1488
.642	.0354197	2715	.2455978	1182	.272179	1484
.643	.0351482	2700	.244386	1180	.270745	1488
.644	.0348782	2686	.243196	1186	.269317	1484
.645	.0346096	2672	.242010	1183	.267893	1420
.646	.0343324	2658	.240827	1180	.265473	1414
.647	.0340766	2643	.239647	1177	.263059	1408
.648	.0338123	2630	.238470	1174	.263650	1405
.649	.0335493	2615	.237296	1171	.262245	1400
.650	.0332878	2602	.236125	1167	.260845	1384
.651	.0330276	2587	.234958	1165	.259451	1381
.652	.0327689	2574	.233793	1161	.258060	1385
.653	.0325115	2560	.232632	1158	.256675	1381
.654	.0322553	2546	.231473	1155	.255294	1378
.655	.0320009	2532	.230318	1152	.253918	1371
.656	.0317477	2518	.229166	1148	.252547	1367
.657	.0314958	2505	.228017	1145	.251180	1361
.658	.0312453	2491	.226871	1143	.249819	1358
.659	.0309962	2478	.225728	1138	.248461	1352
.660	.0307484	2464	.224589	1137	.247109	1348
.661	.0305020	2451	.223452	1134	.245761	1344
.662	.0302569	2438	.222318	1131	.244417	1338
.663	.0300131	2424	.221187	1127	.243079	1334
.664	.0297707	2411	.220060	1125	.241745	1330
.665	.0295296	2397	.218935	1121	.240415	1325
.666	.0292899	2384	.217814	1118	.239090	1321
.667	.0290515	2371	.216695	1115	.237769	1318
.668	.0288114	2358	.215580	1113	.236453	1311
.669	.0285766	2345	.214467	1108	.235142	1307
.670	.0283441	2332	.213358	1107	.233835	1302
.671	.0281109	2318	.212251	1103	.232533	1308
.672	.0278790	2306	.211148	1101	.231235	1294
.673	.0276484	2293	.210047	1087	.229941	1288
.674	.0274191	2280	.208950	1085	.228652	1285
.675	.0271911	2267	.207855	1082	.227367	1280
.676	.0269644	2255	.206763	1088	.226087	1278
.677	.0267389	2241	.205675	1086	.224811	1272
.678	.0265118	2229	.204589	1082	.223539	1267
.679	.0262919	2217	.203507	1080	.222272	1263
.680	.0260702	2204	.202427	1077	.221009	1258
.681	.0258498	2191	.201350	1074	.219751	1254
.682	.0256307	2179	.200276	1071	.218497	1250
.683	.0254128	2166	.199205	1068	.217247	1248
.684	.0251962	2154	.198137	1065	.216001	1241
.685	.0249808	2141	.197072	1062	.214760	1237
.686	.0247667	2128	.196010	1058	.213523	1233
.687	.0245538	2117	.194951	1056	.212290	1228
.688	.0243421	2104	.193895	1054	.211062	1224
.689	.0241317	2093	.192841	1050	.209838	1221
.690	.0239224	2080	.191791	1048	.208617	1215
.691	.0237144	2068	.190743	1044	.207402	1212
.692	.0235076	2055	.189699	1042	.206190	1208
.693	.0233021	2044	.188657	1039	.204982	1203
.694	.0230977	2032	.187618	1036	.203779	1199
.695	.0228945	2020	.186582	1033	.202580	1188
.696	.0226925	2008	.185549	1030	.201385	1181
.697	.0224917	1995	.184519	1028	.200194	1187
.698	.0222921	1984	.183491	1024	.199007	1182
.699	.0220937	1972	.182467	1022	.197825	1179
.700	.0218965		.181445		.196646	
					.895588	
						1.02020

NACA

TABLE I.— CONTINUED

t	a(t)	b(t)	c(t)	d(t)	e(t)	f(t)
.700	.0218965	1861	.181445 1019	.196646 1174	.895588 2000	1.02020 285
.701	.0217004	1849	.180426 1015	.195472 1171	.893588 2000	1.01735 285
.702	.0215055	1837	.179411 1014	.194301 1166	.891588 2001	1.01450 285
.703	.0213116	1825	.178397 1010	.193135 1162	.889587 2000	1.01165 284
.704	.0211193	1814	.177387 1007	.191973 1158	.887587 2000	1.00881 284
.705	.0209279	1803	.176380 1005	.190814 1154	.885587 2000	1.00597 284
.706	.0207376	1891	.175375 1001	.189660 1150	.883587 2000	1.00313 283
.707	.0205485	1879	.174374 999	.188510 1146	.881587 2000	1.00030 283
.708	.0203606	1868	.173375 998	.187364 1143	.879587 2000	.997475 2823
.709	.0201738	1857	.172379 994	.186221 1138	.877587 2000	.994652 2819
.710	.0199881	1845	.171385 990	.185083 1134	.875587 2000	.991833 2815
.711	.0198036	1834	.170395 988	.183949 1130	.873587 2000	.989018 2811
.712	.0196202	1822	.169407 985	.182819 1127	.871587 2000	.986207 2807
.713	.0194380	1811	.168422 981	.181692 1124	.869587 2001	.983400 2804
.714	.0192569	1801	.167441 980	.180570 1118	.867586 2000	.980596 2800
.715	.0190768	1788	.166461 976	.179451 1114	.865586 2001	.977796 2786
.716	.0188979	1777	.165485 974	.178337 1111	.863585 2001	.975000 2782
.717	.0187202	1767	.164511 970	.177226 1107	.861584 2000	.972208 2780
.718	.0185435	1755	.163541 969	.176119 1106	.859584 2001	.969419 2785
.719	.0183679	1744	.162572 965	.175016 1098	.857583 2002	.966634 2781
.720	.0181935	1734	.161607 962	.173917 1088	.855581 2001	.963853 2778
.721	.0180201	1723	.160645 960	.172822 1081	.853580 2002	.961075 2774
.722	.0178478	1712	.159685 957	.171731 1083	.851578 2002	.958301 2771
.723	.0176766	1701	.158728 954	.170643 1084	.849576 2002	.955530 2768
.724	.0175065	1690	.157774 951	.169559 1078	.847574 2002	.952762 2764
.725	.0173375	1678	.156823 948	.168480 1076	.845572 2003	.949998 2760
.726	.0171696	1669	.155874 946	.167404 1073	.843569 2003	.947238 2757
.727	.0170027	1658	.154928 943	.166331 1068	.841566 2004	.944481 2754
.728	.0168369	1648	.153985 940	.165263 1065	.839562 2004	.941727 2751
.729	.0166721	1638	.153045 938	.164198 1061	.837558 2004	.938976 2747
.730	.0165095	1626	.152107 935	.163137 1057	.835554 2004	.936229 2744
.731	.0163459	1616	.151172 932	.162080 1050	.833550 2005	.933485 2741
.732	.0161843	1605	.150240 929	.161027 1050	.831545 2005	.930744 2738
.733	.0160238	1594	.149311 927	.159977 1046	.829539 2005	.928006 2734
.734	.0158644	1584	.148384 924	.158931 1042	.827534 2007	.925272 2732
.735	.0157060	1574	.147460 921	.157889 1038	.825527 2006	.922540 2728
.736	.0155486	1565	.146539 918	.156850 1034	.823521 2008	.919812 2725
.737	.0153923	1555	.145621 915	.155816 1032	.821513 2007	.917086 2722
.738	.0152370	1543	.144705 913	.154784 1027	.819506 2008	.914364 2718
.739	.0150827	1533	.143792 910	.153757 1024	.817497 2008	.911645 2717
.740	.0149294	1522	.142882 908	.152733 1020	.815488 2008	.908928 2713
.741	.0147772	1512	.141974 905	.151713 1016	.813479 2010	.906215 2711
.742	.0146260	1502	.141069 902	.150697 1013	.811469 2011	.903504 2708
.743	.0144758	1492	.140167 900	.149688 1008	.809458 2011	.900796 2705
.744	.0143266	1481	.139268 897	.148675 1005	.807474 2012	.898091 2702
.745	.0141785	1472	.138371 894	.147670 1002	.805435 2012	.895389 2700
.746	.0140313	1462	.137477 892	.146668 898	.803423 2014	.892689 2697
.747	.0138851	1451	.136585 890	.145670 895	.801409 2015	.889992 2694
.748	.0137400	1442	.135567 888	.144675 891	.799396 2015	.887298 2692
.749	.0135958	1432	.134811 883	.143688 897	.797381 2016	.884606 2693
.750	.0134526	1422	.133928 881	.142697 894	.795365 2016	.881917 2690
.751	.0133104	1412	.133047 878	.141713 891	.793349 2017	.879231 2684
.752	.0131692	1402	.132169 875	.140732 876	.791332 2018	.876547 2682
.753	.0130290	1393	.131294 873	.139756 873	.789314 2018	.873865 2678
.754	.0128897	1383	.130421 869	.138783 870	.787296 2020	.871186 2676
.755	.0127514	1373	.129552 868	.137813 865	.785276 2020	.868510 2674
.756	.0126141	1364	.128634 864	.136847 863	.783256 2021	.865836 2672
.757	.0124777	1354	.127820 862	.135884 859	.781235 2022	.863164 2670
.758	.0123423	1345	.126958 859	.134925 855	.779213 2023	.860494 2667
.759	.0122078	1334	.126099 857	.133970 852	.777190 2024	.857827 2665
.760	.0120744		.125242	.133018	.775166	.855162

NACA

TABLE I.- CONTINUED

t	a(t)	b(t)	c(t)	d(t)	e(t)	f(t)
.760	.0120744	1226	.125242	858	.133018	848
.761	.0119418	1215	.124389	852	.132070	845
.762	.0118102	1204	.123537	848	.131125	842
.763	.0116796	1203	.122689	846	.130183	838
.764	.0115498	1207	.121843	843	.129245	834
.765	.0114211	1278	.121000	841	.128311	831
.766	.0112932	1208	.120159	838	.127380	828
.767	.0111663	1200	.119321	835	.126452	824
.768	.0110403	1201	.118486	832	.125528	821
.769	.0109152	1241	.117654	830	.124607	817
.770	.0107911	1232	.116824	828	.123690	814
.771	.0106679	1224	.115996	824	.122776	810
.772	.0105455	1214	.115172	822	.121866	807
.773	.0104241	1205	.114350	820	.120959	803
.774	.0103036	1186	.113530	816	.120056	801
.775	.0101840	1187	.112714	815	.119155	806
.776	.0100653	1178	.111899	811	.118259	804
.777	.0099475	1168	.111088	809	.117365	800
.778	.0098306	1161	.110279	806	.116475	800
.779	.0097145	1151	.109473	804	.115589	800
.780	.0095994	1148	.108669	801	.114706	800
.781	.0094851	1138	.107868	788	.113826	877
.782	.0093718	1126	.107070	786	.112949	873
.783	.0092592	1118	.106274	783	.112076	869
.784	.0091476	1108	.105481	780	.111207	867
.785	.0090368	1098	.104691	788	.110340	853
.786	.0089269	1080	.103903	785	.109477	860
.787	.0088179	1082	.103118	784	.108617	856
.788	.0087097	1074	.102335	780	.107761	853
.789	.0086023	1064	.101555	777	.106908	850
.790	.0084959	1057	.100778	773	.106058	847
.791	.0083902	1048	.100003	772	.105211	843
.792	.0082854	1039	.0992306	7688	.100238	840
.793	.0081815	1031	.0984610	7686	.103328	836
.794	.0080784	1023	.0976941	7644	.102692	833
.795	.0079761	1014	.0969297	7617	.101859	830
.796	.0078747	1007	.0961680	7581	.101029	827
.797	.0077740	997	.0954089	7585	.100223	824
.798	.0076743	989	.0946524	7589	.0993783	822
.799	.0075753	982	.0938985	7513	.0985581	8170
.800	.0074771	973	.0931472	7487	.0977411	8137
.801	.0073798	965	.0923985	7443	.0969274	8104
.802	.0072833	957	.0916524	7435	.0961170	8072
.803	.0071876	949	.0909089	7408	.0953098	8040
.804	.0070927	941	.0901680	7383	.0945058	8007
.805	.0069986	934	.0894297	7357	.0937051	7975
.806	.0069052	925	.0886940	7331	.0929076	7943
.807	.0068127	917	.0879609	7304	.0921133	7910
.808	.0067210	909	.0872303	7278	.0913223	7878
.809	.0066301	901	.0865026	7253	.0905345	7846
.810	.0065400	894	.0857773	7227	.0897499	7815
.811	.0064506	885	.0850546	7201	.0889684	7782
.812	.0063620	878	.0843345	7175	.0881902	7750
.813	.0062742	870	.0836170	7148	.0874152	7718
.814	.0061872	863	.0829021	7123	.0866434	7686
.815	.0061009	855	.0821896	7087	.0858748	7654
.816	.0060154	847	.0814801	7071	.0851094	7623
.817	.0059307	840	.0807730	7045	.0843471	7580
.818	.0058467	832	.0800685	7018	.0835881	7558
.819	.0057635	824	.0793666	6993	.0828322	7527
.820	.0056811		.0786673		.0820795	
					.651031	
					.698004	
					.74714	

NACA

TABLE I.— CONTINUED

t	a(t)	b(t)	c(t)	d(t)	e(t)	f(t)
.820	.0056811	817	.0786673	8867	.0820795	7485
.821	.0055994	810	.0779706	8841	.0813300	7464
.822	.0055184	802	.0772765	8815	.0805836	7432
.823	.0054382	795	.0765850	8888	.0798404	7401
.824	.0053587	787	.0758961	8863	.0791003	7369
.825	.0052800	780	.0752098	8837	.0783634	7358
.826	.0052020	773	.0745261	8811	.0776296	7305
.827	.0051247	765	.0738450	8785	.0768990	7274
.828	.0050482	758	.0731665	8758	.0761716	7243
.829	.0049724	751	.0724906	8733	.0754743	7212
.830	.0048973	744	.0718173	8707	.0747261	7181
.831	.0048229	736	.0711466	8681	.0740080	7149
.832	.0047493	729	.0704785	8655	.0738931	7118
.833	.0046764	723	.0698130	8628	.0725813	7088
.834	.0046041	715	.0691501	8603	.0718727	7055
.835	.0045326	708	.0684898	8577	.0711672	7028
.836	.0044618	701	.0678321	8551	.0704647	6883
.837	.0043917	694	.0671770	8524	.0697654	6861
.838	.0043223	688	.0665246	8498	.0690693	6831
.839	.0042535	680	.0658747	8472	.0683762	6809
.840	.0041855	673	.0652275	8447	.0676863	6888
.841	.0041182	667	.0645928	8420	.066994	6837
.842	.0040515	660	.0639408	8394	.0663157	6806
.843	.0039855	653	.0633014	8368	.0656351	6776
.844	.0039202	646	.0626646	8341	.0649575	6744
.845	.0038556	638	.0620305	8316	.0642831	6713
.846	.0037917	633	.0613989	8289	.0636118	6682
.847	.0037284	622	.0607700	8263	.0629436	6652
.848	.0036658	620	.0601437	8237	.0622784	6620
.849	.0036038	613	.0593200	8211	.0616164	6598
.850	.0035425	606	.0588699	8184	.0609573	6559
.851	.0034819	600	.0582803	8158	.0603016	6528
.852	.0034219	593	.0576647	8132	.0596488	6496
.853	.0033626	587	.0570915	8105	.0589992	6466
.854	.0033039	580	.0564410	8078	.0583526	6435
.855	.0032459	574	.0558331	8053	.0577091	6404
.856	.0031885	567	.0552278	8028	.0570687	6373
.857	.0031318	561	.0546252	8000	.0564314	6343
.858	.0030757	555	.0540252	5873	.0557971	6312
.859	.0030202	548	.0534279	5947	.0551659	6280
.860	.0029553	542	.0528332	5920	.0545379	6250
.861	.0029111	535	.0522412	5894	.0539129	6220
.862	.0028575	530	.0516518	5867	.0532909	6188
.863	.0028045	523	.0510651	5841	.0526721	6158
.864	.0027522	518	.0504810	5814	.0520563	6127
.865	.0027004	511	.0498996	5788	.0514436	6088
.866	.0026493	505	.0493208	5761	.0508340	6065
.867	.0025987	498	.0487447	5734	.0502275	6034
.868	.0025488	493	.0481713	5707	.0496241	6004
.869	.0024995	487	.0476006	5681	.0490237	5979
.870	.0024508	482	.0470325	5654	.0484264	5942
.871	.0024026	475	.0464671	5627	.0478322	5911
.872	.0023551	470	.0459044	5600	.0472411	5881
.873	.0023081	463	.0453444	5573	.0466530	5648
.874	.0022618	458	.0447871	5547	.0460681	5619
.875	.0022160	452	.0442324	5518	.0454862	5788
.876	.0021708	446	.0436805	5492	.0449074	5757
.877	.0021262	441	.0431313	5465	.0443317	5727
.878	.0020821	434	.0425848	5438	.0437590	5685
.879	.0020387	429	.0420409	5411	.0431895	5664
.880	.0019958		.0414998		.0426231	
					.516474	
						.539743
						2.10538

TABLE I.— CONCLUDED

t	a(t)	b(t)	c(t)	d(t)	e(t)	f(t)
.830	.0019958	424	.0414998	5383	.0426231	5634
.831	.0019534	417	.0409615	5557	.0420597	5608
.832	.0019117	413	.0404258	5529	.0414994	5572
.833	.0018704	408	.0396929	5602	.0409422	5640
.834	.0018298	401	.0393627	5675	.0403882	5610
.835	.0017897	386	.0388352	5247	.0398372	5476
.836	.0017501	380	.0383105	5220	.0392893	5448
.837	.0017111	355	.0377885	5182	.0387445	5417
.838	.0016726	378	.0372693	5185	.0382028	5365
.839	.0016347	374	.0367528	5137	.0376642	5354
.840	.0015973	366	.0362391	5106	.0371288	5324
.841	.0015604	365	.0357282	5082	.0365964	5283
.842	.0015241	358	.0352200	5053	.0360671	5251
.843	.0014883	358	.0347147	5086	.0355410	5280
.844	.0014530	348	.0342121	4988	.0350180	5189
.845	.0014182	342	.0337123	4970	.0344951	5107
.846	.0013840	337	.0332153	4942	.0339814	5137
.847	.0013503	332	.0327211	4914	.0334677	5105
.848	.0013171	327	.0322297	4885	.0329572	5073
.849	.0012844	322	.0317412	4858	.0324499	5042
.850	.0012522	317	.0312554	4828	.0319457	5011
.851	.0012205	312	.0307725	4801	.0314446	4878
.852	.0011893	307	.0302924	4772	.0309467	4848
.853	.0011586	302	.0298152	4745	.0304519	4818
.854	.0011284	298	.0293409	4715	.0299603	4844
.855	.0010986	292	.0288694	4687	.0294719	4853
.856	.0010694	287	.0284007	4657	.0289866	4820
.857	.0010407	283	.0279350	4628	.0285046	4780
.858	.0010124	278	.0274721	4600	.0280256	4757
.859	.0009846	273	.0270121	4570	.0275499	4725
.860	.0009573	268	.0265551	4542	.0270774	4693
.861	.0009305	264	.0261009	4512	.0266081	4661
.862	.0009041	258	.0256497	4483	.0261420	4628
.863	.0008782	255	.0252014	4454	.0256791	4597
.864	.0008527	250	.0247560	4424	.0252194	4564
.865	.0008277	245	.0243136	4395	.0247630	4532
.866	.0008032	241	.0238741	4364	.0243098	4500
.867	.0007791	236	.0234377	4335	.0238598	4467
.868	.0007555	232	.0230042	4305	.0234131	4425
.869	.0007323	228	.0225737	4275	.0229696	4402
.870	.0007095	223	.0221462	4245	.0225294	4388
.871	.0006872	218	.0217217	4215	.0220925	4338
.872	.0006654	215	.0213002	4184	.0216589	4308
.873	.0006439	210	.0208818	4154	.0212286	4271
.874	.0006229	206	.0204664	4123	.0208015	4257
.875	.0006023	202	.0200541	4082	.0203778	4204
.876	.0005821	197	.0196149	4052	.0199574	4170
.877	.0005621	193	.0192387	4030	.0195404	4137
.878	.0005431	180	.0188357	4000	.0191267	4104
.879	.0005241	185	.0184357	3968	.0187163	4068
.880	.0005056	181	.0180389	3935	.0183094	4036
.881	.0004875	177	.0176453	3905	.0179058	4002
.882	.0004698	173	.0172548	3873	.0175056	3988
.883	.0004525	169	.0168675	3841	.0171088	3934
.884	.0004356	165	.0164834	3808	.0167154	3898
.885	.0004191	162	.0161025	3777	.0163255	3865
.886	.0004029	157	.0157248	3745	.0159390	3880
.887	.0003872	154	.0153503	3711	.0155560	3798
.888	.0003718	150	.0149792	3680	.0151764	3760
.889	.0003568	146	.0146112	3648	.0148004	3725
.890	.0003422		.0142466		.0144279	

















TABLE II.—CONTINUED

t	g(t)	h(t)	i(t)	j(t)	k(t)	l(t)	m(t)	
.580	.112628	580	.529392	1885	.621464	2282	1.21364	70
.581	.112008	581	.527757	1882	.619182	2275	1.21294	71
.582	.111390	582	.526125	1880	.616907	2269	1.21223	68
.583	.110774	583	.524495	1887	.614638	2263	1.21154	70
.584	.110161	584	.522868	1884	.612375	2256	1.21084	70
.585	.109549	585	.521244	1882	.610119	2250	1.21014	68
.586	.108940	586	.519622	1880	.507869	2244	1.20945	70
.587	.108334	587	.518002	1818	.506265	2237	1.20875	68
.588	.107729	588	.516386	1814	.503388	2231	1.20806	69
.589	.107127	589	.514772	1812	.501157	2225	1.20737	68
.590	.106527	590	.513160	1808	.500032	2218	1.20668	68
.591	.105929	591	.511551	1807	.500032	2213	1.20599	68
.592	.105333	592	.509944	1804	.500030	2206	1.20530	68
.593	.104740	593	.508340	1801	.500034	2200	1.20462	68
.594	.104149	594	.506739	1598	.500094	2195	1.20393	68
.595	.103560	595	.505140	1587	.500099	2188	1.20325	68
.596	.102973	596	.503543	1584	.500071	2182	1.20257	68
.597	.102388	597	.501949	1581	.500029	2177	1.20189	68
.598	.101806	598	.500358	1580	.500032	2170	1.20121	68
.599	.101226	599	.498768	1580	.500009	2165	1.20053	68
.600	.100648	600	.497182	1584	.500017	2158	1.19985	67
.601	.100072	601	.495598	1582	.500059	2153	1.19918	68
.602	.0994979	602	.494016	1578	.500006	2147	1.19850	67
.603	.0989263	603	.492437	1577	.500059	2141	1.19783	67
.604	.0983568	604	.490860	1575	.500018	2135	1.19716	67
.605	.0977894	605	.489205	1572	.500028	2130	1.19649	67
.606	.0972242	606	.487713	1570	.500013	2124	1.19582	67
.607	.0966611	607	.486143	1567	.500029	2118	1.19515	67
.608	.0961002	608	.484576	1565	.500011	2113	1.19448	68
.609	.0955413	609	.483011	1562	.500078	2107	1.19382	67
.610	.0949846	610	.481449	1561	.500061	2101	1.19315	68
.611	.0944299	611	.479888	1558	.500050	2096	1.19249	68
.612	.0938774	612	.478330	1555	.500044	2090	1.19183	68
.613	.0933269	613	.476775	1553	.500004	2085	1.19117	68
.614	.0927786	614	.475222	1551	.500019	2079	1.19051	68
.615	.0922323	615	.473671	1548	.500040	2073	1.18985	68
.616	.0916881	616	.472123	1546	.500067	2069	1.18919	68
.617	.091160	617	.470577	1544	.500098	2063	1.18854	68
.618	.0906059	618	.469033	1542	.500035	2057	1.18788	68
.619	.0900079	619	.467491	1539	.500078	2052	1.18723	68
.620	.0895319	620	.465952	1537	.500026	2047	1.18658	68
.621	.0889980	621	.464415	1535	.5000879	2041	1.18593	68
.622	.0884660	622	.462880	1532	.5000838	2036	1.18528	68
.623	.0879362	623	.461348	1530	.5000802	2031	1.18463	68
.624	.0874085	624	.459818	1528	.5000771	2025	1.18398	68
.625	.0868828	625	.458290	1525	.500046	2021	1.18333	68
.626	.0863391	626	.456765	1524	.5000725	2015	1.18269	68
.627	.0858373	627	.455241	1521	.5000710	2010	1.18204	68
.628	.0853176	628	.453720	1519	.5000700	2005	1.18140	68
.629	.0847999	629	.452201	1516	.5000695	1999	1.18076	68
.630	.0842842	630	.450605	1515	.500065	1995	1.18012	68
.631	.0837705	631	.449170	1512	.5000701	1989	1.17948	68
.632	.0832987	632	.447658	1510	.5000712	1985	1.17884	68
.633	.0827490	633	.446148	1508	.500087	1979	1.17820	68
.634	.0822413	634	.444640	1506	.5000748	1974	1.17757	68
.635	.0817355	635	.443134	1503	.500074	1970	1.17693	68
.636	.0812317	636	.441631	1501	.5000804	1964	1.17630	68
.637	.0807299	637	.440130	1499	.5000840	1960	1.17567	68
.638	.0802301	638	.438631	1497	.5000880	1954	1.17503	68
.639	.0797322	639	.437134	1495	.5000926	1950	1.17440	68
.640	.0792363	640	.435639	1492	.5000976	1944	1.17378	68

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TABLE II.— CONTINUED

t	g(t)	h(t)	i(t)	j(t)	k(t)	l(t)	m(t)
.640	.0792363	4840	.435639	1482	.494976	1944	1.17378
.641	.0787423	4821	.434147	1481	.493032	1840	1.17315
.642	.0782502	4801	.432656	1482	.491092	1888	1.17252
.643	.0777601	4883	.431168	1480	.489157	1881	1.17189
.644	.0772718	4882	.429682	1485	.487226	1828	1.17127
.645	.0767856	4844	.428197	1482	.485301	1821	1.17064
.646	.0763012	4824	.426715	1478	.483380	1816	1.17002
.647	.0758188	4805	.425236	1478	.481461	1911	1.16940
.648	.0753383	4788	.423758	1478	.479553	1806	1.16878
.649	.0748597	4767	.422282	1478	.477647	1802	1.16816
.650	.0743830	4748	.420809	1472	.475745	1887	1.16754
.651	.0739082	4728	.419337	1483	.473848	1863	1.16692
.652	.0734354	4710	.417868	1467	.471955	1888	1.16631
.653	.0729644	4691	.416401	1488	.470067	1883	1.16569
.654	.0724953	4673	.414935	1488	.468184	1876	1.16508
.655	.0720280	4654	.413472	1481	.466305	1874	1.16447
.656	.0715626	4635	.412011	1459	.464431	1863	1.16385
.657	.0710991	4616	.410552	1457	.462562	1885	1.16324
.658	.0706375	4588	.409095	1455	.460697	1881	1.16263
.659	.0701777	4578	.407640	1455	.458836	1888	1.16202
.660	.0697199	4551	.406187	1451	.456980	1851	1.16142
.661	.0692638	4541	.404736	1448	.455129	1848	1.16081
.662	.0688097	4525	.403287	1447	.453261	1842	1.16020
.663	.0683572	4505	.401840	1445	.451439	1838	1.15960
.664	.0679067	4487	.400395	1443	.449600	1833	1.15900
.665	.0674580	4468	.398052	1441	.447767	1830	1.15839
.666	.0670112	4450	.397511	1438	.445937	1825	1.15779
.667	.0665662	4432	.396072	1427	.444112	1821	1.15719
.668	.0661230	4414	.394635	1435	.442291	1817	1.15659
.669	.0656816	4398	.393200	1433	.440474	1812	1.15599
.670	.0652420	4378	.391767	1432	.438666	1808	1.15540
.671	.0648042	4359	.390335	1428	.436854	1804	1.15480
.672	.0643683	4342	.388906	1427	.435050	1798	1.15420
.673	.0639341	4323	.387479	1425	.433251	1785	1.15361
.674	.0635018	4306	.386054	1424	.431456	1781	1.15302
.675	.0630712	4287	.384630	1421	.429665	1787	1.15242
.676	.0626425	4270	.383209	1418	.427878	1783	1.15183
.677	.0622155	4252	.381790	1418	.426095	1778	1.15124
.678	.0617903	4233	.380372	1416	.424316	1774	1.15065
.679	.0613568	4216	.378956	1414	.422542	1770	1.15006
.680	.0609424	4198	.377543	1418	.420772	1788	1.14948
.681	.0605233	4181	.376131	1410	.419006	1763	1.14889
.682	.0601072	4164	.374721	1408	.417243	1758	1.14830
.683	.0596908	4146	.373313	1406	.415485	1754	1.14772
.684	.0592762	4128	.371907	1405	.413731	1750	1.14714
.685	.0588634	4112	.370502	1402	.411981	1748	1.14655
.686	.0584522	4093	.369100	1401	.410235	1742	1.14597
.687	.0580429	4075	.367699	1388	.408493	1738	1.14539
.688	.0576353	4058	.366301	1387	.406755	1734	1.14481
.689	.0572294	4042	.364904	1388	.405021	1730	1.14423
.690	.0568252	4024	.363509	1388	.403291	1726	1.14365
.691	.0564228	4007	.362116	1382	.401565	1722	1.14308
.692	.0560221	3980	.360724	1388	.399843	1718	1.14250
.693	.0556231	3973	.359335	1388	.398124	1714	1.14193
.694	.0552228	3955	.357947	1388	.396410	1711	1.14135
.695	.0548303	3838	.356361	1384	.394599	1708	1.14078
.696	.0544364	3821	.355177	1388	.392993	1708	1.14021
.697	.0540443	3804	.353795	1380	.391290	1700	1.13964
.698	.0536539	3888	.352415	1378	.389590	1685	1.13907
.699	.0532651	3870	.351036	1378	.387895	1681	1.13850
.700	.0528781	3854	.349660	1378	.386004	1688	1.13793

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TABLE II.— CONTINUED

t	g(t)	h(t)	i(t)	j(t)	k(t)	l(t)	m(t)
.700	.0528781	8854	.349660	1875	.386204	1888	1.13793 57
.701	.0524927	8837	.348265	1874	.384516	1884	1.13736 57
.702	.0521090	8820	.346911	1871	.382832	1880	1.13679 56
.703	.0517270	8803	.345540	1870	.381152	1877	1.13623 57
.704	.0513467	8786	.344170	1867	.379475	1875	1.13566 56
.705	.0509681	8770	.342803	1866	.377802	1889	1.13510 56
.706	.0505911	8753	.341437	1865	.376133	1865	1.13454 56
.707	.0502158	8736	.340072	1862	.374468	1862	1.13398 57
.708	.0498422	8720	.338710	1861	.372806	1858	1.13341 56
.709	.0494702	8703	.337349	1859	.371148	1855	1.13285 55
.710	.0490999	8687	.335990	1857	.369493	1851	1.13230 56
.711	.0487312	8670	.334633	1856	.367842	1847	1.13174 56
.712	.0483642	8654	.333277	1854	.366195	1844	1.13118 56
.713	.0479988	8637	.331923	1852	.364551	1840	1.13062 55
.714	.0476351	8621	.330571	1850	.362911	1837	1.13007 56
.715	.0472730	8605	.329221	1848	.361274	1835	1.12951 55
.716	.0469125	8588	.327872	1847	.359641	1829	1.12896 55
.717	.0465537	8572	.326525	1845	.358012	1826	1.12841 56
.718	.0461965	8556	.325180	1844	.356386	1822	1.12785 55
.719	.0458409	8539	.323836	1841	.354764	1819	1.12730 55
.720	.0454870	8523	.322495	1841	.353145	1818	1.12675 55
.721	.0451347	8508	.321154	1838	.351529	1812	1.12620 55
.722	.0447839	8491	.319816	1837	.349917	1808	1.12565 54
.723	.0444348	8475	.318479	1835	.348308	1805	1.12511 55
.724	.0440873	8459	.317144	1833	.346703	1801	1.12456 55
.725	.0437414	8443	.315811	1832	.345102	1800	1.12401 54
.726	.0433971	8427	.314479	1830	.343503	1805	1.12347 55
.727	.0430544	8411	.313149	1828	.341908	1591	1.12292 54
.728	.0427133	8395	.311821	1827	.340317	1588	1.12238 54
.729	.0423738	8380	.310494	1825	.338729	1585	1.12184 54
.730	.0420358	8363	.309169	1824	.337144	1582	1.12130 55
.731	.0416995	8348	.307845	1821	.335562	1578	1.12075 54
.732	.0413647	8332	.306524	1821	.333984	1575	1.12021 54
.733	.0410315	8316	.305203	1818	.332409	1571	1.11967 53
.734	.0406999	8301	.303885	1817	.330838	1568	1.11914 54
.735	.0403698	8284	.302568	1815	.329270	1565	1.11860 54
.736	.0400414	8270	.301253	1814	.327705	1562	1.11806 53
.737	.0397144	8253	.299939	1812	.326143	1558	1.11753 54
.738	.0393891	8238	.298627	1810	.324585	1556	1.11699 53
.739	.0390653	8223	.297317	1808	.323029	1552	1.11646 54
.740	.0387430	8207	.296008	1807	.321477	1548	1.11592 53
.741	.0384223	8192	.294701	1806	.319929	1546	1.11539 53
.742	.0381031	8176	.293395	1804	.318383	1542	1.11486 53
.743	.0377855	8160	.292091	1802	.316841	1539	1.11433 53
.744	.0374695	8145	.290789	1801	.315302	1536	1.11380 53
.745	.0371549	8130	.289488	1298	.313766	1533	1.11327 53
.746	.0368419	8114	.288189	1298	.312233	1530	1.11274 53
.747	.0365305	8100	.286891	1295	.310703	1527	1.11221 52
.748	.0362205	8084	.285595	1293	.309176	1523	1.11169 53
.749	.0359121	8068	.284300	1293	.307653	1520	1.11116 52
.750	.0356052	8054	.283007	1291	.306133	1518	1.11064 53
.751	.0352998	8038	.281716	1290	.304615	1514	1.11011 52
.752	.0349960	8024	.280426	1288	.303101	1511	1.10959 53
.753	.0346936	8008	.279138	1287	.301590	1508	1.10906 53
.754	.0343928	2993	.277851	1285	.300082	1505	1.10854 52
.755	.0340935	2979	.276566	1284	.298577	1502	1.10802 52
.756	.0337956	2963	.275282	1282	.297075	1498	1.10750 52
.757	.0334993	2948	.274000	1280	.295576	1496	1.10698 52
.758	.0332045	2933	.272720	1280	.294080	1493	1.10646 52
.759	.0329112	2919	.271440	1277	.292587	1490	1.10594 51
.760	.0326193	2904	.270163	1275	.291097	1487	1.10543 52
						1.27677	152
						.159898	181
						.516863	782

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th	$a(t)$			
4m	(19) - [All V's above]			
4n	$\frac{4o}{4m} \times 4g$			
4s	$\frac{4m}{4n} \div 4h$			
4t	$4s \times 4o \times 4g \times 4e$			
4u	$4s \times 4o \times 4f$			
4v	$4s \times 4o \times 4g$			
4w	$4s \times 4h$			
5a	(19) - (18) from column $P_4$	4]		
5d	$17 \div 5a$			
5e	$a(t)$			
5f	$b(t)$	From Table I		
5g	$c(t)$	as functions of (5d)		
5h	$d(t)$			
5m	(19) - [All V's above]			
5n	$5a \times 5g$			
5s	$5m \div 5n$			
5t	$5s \times 5o \times 5a \times 5e$			
5u	$5s \times 5o \times 5f$			
5v	$5s \times 5a \times 5g$			
5w	$5s \times 5h$			
6a	(19) - (18) from column $P_5$	4]		
6d	$17 \div 6a$			
6e	$a(t)$			
6f	$b(t)$	From Table I		
6g	$c(t)$	as functions of (6d)		
6h	$d(t)$			
6m	(19) - [All V's above]			
6n	$6a \times 6g$			
6s	$6m \div 6n$			
6t	$6s \times 6a \times 6a \times 6e$			
6u	$6s \times 6a \times 6f$			
6v	$6s \times 6a \times 6g$			
6w	$6s \times 6h$			
20	Add all t's			
21	Add all U's			
22	Add all V's			
23	Add all W's			
24	$5 \times 23$			
25	$24 + 14$			
26	$4 \times 23$			
27	$23 + 26$			
28	$15 \times 27$			
29	$16 + 28$			
30	$8 \times 14$			
31	$50 \times 24$			
32	$30 \times 28$			
33	$30 \times 27$			
34	$31 - 20$			
35	$32 - 21$			
36	$33 + 1 \times 24$			
37	$33 - 36$			
38	$3 \times 14 \times 28$			
39	$3 \times 14 \times 27$			
40	$33 - 24$			
41	$\frac{1}{2} \times 24 \times 24$			
42	$23 \times 34$			
43	$21 \times 35$			
44	$38 \times 41$			
45	$42 + 43 + 44$			
46	$28 \times 34$			
47	$21 \times 37$			
48	$40 \times 41$			
49	$46 + 47 + 48$			
50	$21 \times 34$			
51	$14 \times 24 \times 41$			
52	$60 + 51$			

4mm	(60) - [All VV's above]			
4ss	$(4mm) \div (4m)$			
4uu	$(4ss) \times (4u)$			
4vv	$(4ss) \times (4v)$			
5mm	(60) - [All VV's above]			
5ss	$(5mm) \div (5m)$			
5uu	$(5ss) \times (5u)$			
5vv	$(5ss) \times (5v)$			
6mm	(60) - [All VV's above]			
6ss	$(6mm) \div (6m)$			
6uu	$(6ss) \times (6u)$			
6vv	$(6ss) \times (6v)$			
61	Check: (6a) should equal (13)			
62	Check: (62) should equal (6)			
63	1 - (61)			
64	$(63) \times (63)$			
65	$(4) \times (62) \times (62)$			
66	1 - (64) - (65)			
67	$(10) \times (66)$			
68	1 + (67)			
69	$\log_{10}(68)$			
70	$(11) \times (69)$			
71	antilog (70)			
72	(71) - 1			
73	$(12) \times (72)$			
74	Second-order $O_p$			
75	Keep only 3 sig. f			
76	$1 - (74) - (75)$			
77	$(10) \times (76)$			
78	1 + (77)			
79	$\log_{10}(78)$			
80	$(11) \times (79)$			
81	antilog (80)			
82	(81) - 1			
83	$(12) \times (82)$			
First-order $O_p$				
Calculate only on each side of every corner (that column which has a (Cs) somewhere above, and the c				

compute  
fig. figs.  
7 dec.

5	7(4)				
6	1 + (5)				
7	(2) + 1				
8	5(3) x 7 ÷ 4				

Interpolate  
linearly  
in tables

9	(2) - 1				
10	7(2) x 3				
11	(2) ÷ 3				
12	2 ÷ 3 x 3				

**FORM A: Calculation of 2nd-Order Super-sonic Flow Past Body of Revolution**

$P_1$	$P_2$	$P_3$	$P_4$	$P_5$	$P_6$

$P_0$	$P_1$	$P_2$	$P_3$	$P_4$	$P_5$	$P_6$
53	Copy (13)					
54	Copy (13)					
55	1 - (21)					
56	(3) x (45)					
57	(5) x (6) x (45)					
58	(3) x (52)					
59	(54) x (55)					
60	(59) - (57)					

OSS (60) ÷ (54)							
0uu (OSS) x (OU)							
Ovv (OSS) x (OV)							

1mm (60) - [All VV's above]							
1ss (1mm) ÷ (1m)							
1uu (33) x (1v)							
1vv (45) x (1v)							

2mm (60) - [All VV's above]						
2ss (2mm) ÷ (2m)						
2uu (25) x (2u)						
2vv (25) x (2v)						

X

3mm (60) - [All VV's above]						
3ss (3mm) ÷ (3m)						
3uu (33) x (3u)						
3vv (33) x (3v)						

$P_2 \uparrow$   
of 3d  
 $P_3 \uparrow$   
of 4d  
 $P_4 \uparrow$   
of 4d

1mm	(60) - [All VV's above]						
155	(1mm) $\div$ (1m)						
1uu	(155) $\times$ (1u)						
1vv	(155) $\times$ (1v)						

2mm	(60) - [All VV's above]						
255	(2mm) $\div$ (2m)						
2uu	(255) $\times$ (2u)						
2vv	(255) $\times$ (2v)						

3mm	(60) - [All VV's above]						
355	(3mm) $\div$ (3m)						
3uu	(355) $\times$ (3u)						
3vv	(355) $\times$ (3v)						

4mm	(60) - [All VV's above]						
455	(4mm) $\div$ (4m)						
4uu	(455) $\times$ (4u)						
4vv	(455) $\times$ (4v)						

5mm	(60) - [All VV's above]						
555	(5mm) $\div$ (5m)						
5uu	(555) $\times$ (5u)						
5vv	(555) $\times$ (5v)						

6mm	(60) - [All VV's above]						
655	(6mm) $\div$ (6m)						
6uu	(655) $\times$ (6u)						
6vv	(655) $\times$ (6v)						

Check: (22) should equal (1)							
61 (36) + All UU's							
62 (37) + All VV's							

Check: (82) should equal (58)

63	1 - (6)						
64	(63) $\times$ (63)						
65	(4) $\times$ (62) $\times$ (62)						
66	1 - (64) - (65)						
67	(10) $\times$ (66)						
68	1 + (67)						
69	log <sub>10</sub> (68)						
70	(11) $\times$ (68)						
71	antilog (70)						
72	(71) - 1						
73	(12) $\times$ (72)						

Second-order Op

Keep only 3 sig. figs. in final results

74	(62) $\times$ (55)						
75	(13) $\times$ (15)						
76	1 - (74) - (75)						
77	(10) $\times$ (76)						
78	1 + (77)						
79	log <sub>10</sub> (78)						
80	(11) $\times$ (79)						
81	antilog (80)						
82	(81) - 1						
83	(12) $\times$ (82)						

First-order Op

Calculate only on each side of every corner (that is, only for every column which has a (G) somewhere above, and the column preceding it).

## FORM B: Insert at Corner or Curvature Discontinuity

Ca	(13) - [18] from this col. $\rightarrow$
Cb	$\frac{1}{C_a} \times C_b$
Cc	$C_a \times C_c$
Cd	$\frac{C_d}{C_a} \div C_d$
Ce	$\frac{h(t)}{C_d}$
Cf	j(t) From Table II as functions of $C_d$
Cg	$k(t)$
Ch	$l(t)$
Ci	$m(t)$
Cm	(19) - [All V's above]
CS	$C_b \times C_m$
Ct	$C_g \times C_e \times C_b$
Cu	$C_g \times C_f \div C_b$
Cv	$C_g \times C_g \div C_b$
Cw	$- C_s \times C_h \div C_c$
Cx	$C_s \times C_i \div C_c$
Ka	(13) - [18] from this col. $\rightarrow$
Kb	$\frac{1}{C_a} \times C_b$
Kd	$\frac{1}{C_d} \div C_a$
Ke	$g(t)$
Kf	From Table II h(t) as functions of $C_d$
Kg	$i(t)$
Kh	$j(t)$
Kl	$k(t)$
Kj	$3 \times (5) \times C_w$
Kk	$7 \times (4) \times C_w$
Kl	[27] from this col. $\nabla$ - (Kk)
Km	(15) x (Kl)
Kh	[29] from this col. $\nabla$ - (Kj)
Kp	(26) + Km - Kn
Kq	(4) x (15)
Kr	(5) - Kq
Ks	$K_b \times K_p \div K_f$
Kt	$K_s \times K_a \times K_b \times K_e$
Ku	$K_s \times K_b \times K_f$
Kv	$K_b \times K_q$
Kw	$K_s \times K_h \div K_b$
Kx	$K_s \times K_L \div K_b$

Spp	(58) from this $\nabla$ col. $\rightarrow$
Sqr	58 from this col. $\rightarrow$
Srr	$S_{pp} - S_{qq}$
Sss	$S_{rr} \div$ [First Cu]
Suu	$S_{ss} \times C_w$
Svv	$S_{ss} \times C_x$

 $\rightarrow 0$  if no previous column

↑ ↑ 0 if no corner (No Cw ten rows above)

↑ ↑ -{ -Kk } if no previous column

Cmm(60) - [All VV's above]

✓ Omit these 3 rows if no corner  
(no S's directly above)

Css	$C_{mm} \div C_m$
Kuu	$(C_{ss}) \times (K_w)$
Kvv	$(C_{ss}) \times (K_v)$